

EARTH-BASED TRACKING OF A LUNAR BEACON

GPO PRICE \$ _____

CFSTI PRICE(S) \$ _____

Hard copy (HC) 3.00

Microfiche (MF) .75

ff 853 July 85

N66 38707

FACILITY FORM 802	ACCESSION NUMBER	(THRU)
	<u>68</u>	<u>1</u>
	(PAGES)	(CODE)
	<u>CR-78512</u>	<u>07</u>
	(NASA CR OR TMX OR AD NUMBER)	(CATEGORY)

PHILCO. | **WDL DIVISION**
A SUBSIDIARY OF *Ford Motor Company* | 3875 FABIAN WAY, PALO ALTO, CALIFORNIA

WDL-TR3007
19 AUGUST 1966

EARTH-BASED TRACKING OF A LUNAR BEACON

Prepared by

PHILCO CORPORATION
A Subsidiary of Ford Motor Company
WDL Division
Palo Alto, California

This is an experimental
manual.

Contract No. NAS5-9939

Prepared for

GODDARD SPACE FLIGHT CENTER
Greenbelt, Maryland

FOREWORD

This report is written by Philco WDL in partial fulfillment of Contract NAS5-9939 for Goddard Space Flight Center.

The program decks for the program described by this report form another partial fulfillment of Contract NAS5-9939. They have been delivered previously.

TABLE OF CONTENTS

<u>Section</u>		<u>Page</u>
	SUMMARY	v
1	INTRODUCTION	1-1
2	THEORETICAL DESCRIPTION	2-1
3	USE OF THE EBTLB PROGRAM	3-1
4	OUTPUT FROM THE EBTLB PROGRAM	4-1

APPENDICES

A	BRIEF DESCRIPTIONS OF SUBROUTINES USED IN THE EBTLB PROGRAM	A-1
B	NEW SUBROUTINE WRITE-UPS	B-1
	REFERENCES	

SUMMARY

A digital computer program for ascertaining the accuracy of lunar beacon location determination is described. The program simulates tracking of a beacon up to 12 Earth-based tracking stations at one time with capabilities for including the effects of a variety of random and bias errors in the determination. Errors are propagated according to the Schmidt-Kalman filter theory.

SECTION 1

INTRODUCTION

The use of beacons on the Moon's surface as aids in space navigation appears likely in the near future. This likelihood raises the question "How accurately could the location of a lunar beacon be determined by tracking the beacon from Earth-based stations?" A tool for answering this question has been developed--the EBTLB computer program.

The EBTLB (Earth-Based Tracking of a Lunar Beacon) program has been developed by Philco WDL for Goddard Space Flight Center under Contract NAS5-9939. It was originally planned that the capability of the EBTLB program would be included as an additional option of the Mark II Error Propagation Program (Reference 1) delivered to Goddard Space Flight Center by Philco WDL under Contract NAS5-9700. The options of that program were already so numerous and the size so large that it was decided to make EBTLB a separate program, since the nature of the problem was different anyway. The resulting special-purpose program runs rapidly and is small enough to require no linking (space-sharing) on the IBM 7094 computer. The rapid development of the EBTLB program was made possible by utilization of many subroutines and much logic of the Mark II and Patched Conic error propagation programs.

SECTION 2

THEORETICAL DESCRIPTION

The location of the beacon is described by the beacon's selenographic latitude, longitude, and altitude and the beacon's position vector relative to the lunar center is given by

$$R_{\text{selenographic}} = (\text{radius}_{\text{Moon}} + \text{altitude}) \begin{bmatrix} \cos(\text{latitude})\cos(\text{longitude}) \\ \cos(\text{latitude})\sin(\text{longitude}) \\ \sin(\text{latitude}) \end{bmatrix} \quad (1-1)$$

Unit vectors north(N), east(E), and down(D) at the beacon's location may be constructed as functions of the selenographic latitude and longitude. The uncertainty in the beacon's position is formulated as position deviations from $R_{\text{selenographic}}$ along N, E and D.

$$\Delta = \epsilon_n N + \epsilon_e E + \epsilon_d D \quad (1-2)$$

The state vector for this program is taken to be

$$x = \begin{bmatrix} \epsilon_n \\ \epsilon_e \\ \epsilon_d \end{bmatrix} \quad (1-3)$$

This state vector contrasts with the spacecraft position and velocity state vector of six components encountered in the Mark II Error Propagation Program (Reference 1) because a beacon is being observed rather than an orbiting spacecraft. The beacon's state vector consists of three elements which are constant in time so that the state transition matrix is an identity matrix and the state covariance matrix does not change between observations.

The state covariance matrix,

$$P = E(XX^T), \quad (1-4)$$

changes at each observation according to the equation

$$P_{\text{after}} = P_{\text{before}} - P_{\text{before}} H^T (H P_{\text{before}} H + Q)^{-1} H P_{\text{before}} \quad (1-5)$$

where H is the sensitivity of the measurements to state deviations and where Q is the covariance matrix of the random errors in the measurements. Reference 1, "User's Manual for the Mark II Error Propagation Program", contains a more completed description of the "filtering" process in its Section 2. The measurement partials, H , for the station measurements (EBTLB has all of the station measurement capability of the Mark II program) are found in Appendix A of Reference 1 with the provision that the spacecraft's position, R , and velocity, V , are replaced by the beacon's position and velocity relative to the center of the Earth.

$$R = R_{\text{Earth-to-Moon}} + T_{\text{selenographic-to-EEDate}}^R R_{\text{selenographic}} \quad (1-6)$$

$$V = V_{\text{Moon wrt Earth}} + T_{\text{selenographic-to-EEDate}} (\Omega_M \times R_{\text{selenographic}}) \quad (1-7)$$

In equations (1-6) and (1-7), $T_{\text{selenographic-to-EEDate}}$ is the transformation of coordinates from the selenographic system to the Earth's true equator and equinox of date coordinate system and Ω_M is the Moon's angular velocity vector. $R_{\text{Earth-to-Moon}}$ and $V_{\text{Moon wrt Earth}}$ are the Moon's position and velocity relative to the center of the Earth--obtained from the ephemeris tape and rotated into true equator and equinox of date coordinates. It was not known how to treat errors in the ephemeris of the Moon or in the transformations to equator and equinox of date, so these are assumed to be perfectly known in the EBTLB program.

SECTION 3

USE OF THE EBTLB PROGRAM

This section describes the means for running the EBTLB program. The specification of station measurement options is identical to that of the Mark II or Patched Conic error propagation programs and is described fully in Reference 1. Program decks have been delivered to Goddard Space Flight Center for those subroutines unique to the EBTLB program. The remaining required subroutines are to be found in the deck for the Patched Conic error propagation program, delivered earlier to Goddard Space Flight Center.

A. Input

Input to the EBTLB program is of four types: 1) the case header card, 2) floating point data for Common storage in the C array, 3) the control times card, and 4) processing options and changes. Input to EBTLB program is the same as input to the Patched Conic Error Propagation Program except for 2), floating point data for Common storage in the C array.

1. Case Header Card

The first card of each case is the case header card. This card reads in the ISW and BCD arrays according to format (6I1, 11A6).

The ISW Array

Only ISW(1) is used by the program, and for only one reason: ISW(1) is set 1 to stop the run. A one in column # 1 of this card should never appear on a case to be executed. This option is normally used on the last data card in the deck, following the last data card in the last case executed.

This card is printed out before the program stops, so it might be as follows

Col 1 Col 7
1 NO MORE CASES

The BCD Array

Columns 7 through 72 are available for up to 11 words of alphanumeric data describing the case. This data is written out as the case header.

2. Floating Point Data Stored in the INPCOM Common C Array

All data of this type is read in according to format 4(I3,E12.8). That is, from one to four sets of an integer K and a number X to be stored as C(K) = X.

Each case requires data sufficient to define the starting epoch, the lunar beacon, and the initial 3x3 P covariance matrix.

The starting epoch is defined by

C(99) starting time from reference epoch, DH.MS
C(102) starting date YR,MONTH.DAY
C(103) starting time of day HR,MIN,SEC
C(101) maximum time interval of interest for this case,
DH.MS.

Selenographic coordinates of the lunar beacon to be tracked are loaded into C(106-108)

C(106) latitude, degrees
C(107) longitude, degrees
C(108) altitude, meters

The program will stop if $80 < |C(107)| < 280$.

The initial 3x3 covariance matrix is defined by the PI array dimensioned 7 and loaded into C(112) through C(118). PI(1) has no significance, while PI(2-7) are eventually loaded into the P matrix as follows

PI(2)→P(1,1)
PI(3)→P(2,2)
PI(4)→P(3,3)
PI(5)→P(1,2) and P(2,1)
PI(6)→P(1,3) and P(3,1)
PI(7)→P(2,3) and P(3,2)

Thus PI(2-7) must define the initial covariance matrix of the error in the estimate of the beacon location (north-east-down) coordinates.

Note: Unlike the Patched Conic Program the prediction and guidance options have no significance in the EBTLB program.

All data read into the C array by the first case remains in core for subsequent cases, until new data is specified. Thus, if multiple cases are to be run on the same beacon, with the same covariance matrix, only a single blank card is required as input. A blank card signals the end of floating point data to be stored in the INPCOM Common array.

3. Control Times Card

See (Reference 1).

4. Processing Options and Changes

Method of inputting these two types of data is identical to the Mark II Error Propagation Program and is described in great detail in that program's User's Manual under "Specification of Error Source Input Data".

Note: This program requires only station data, since the equation of motion errors, vehicle onboard and multiple beacon measurement options have no application in the EBTLB Program.

B. Deck and Tape Setup

The running deck requires 9 binary decks unique to the EBTLB program, 3 decks which also are used by the PINT portion of the Mark II Program, and 21 additional routines in the form of binary decks or 21 IBLDR cards with appropriate IEDIT cards to read the Patched Conic Error Propagation Program's IEDIT tape. (Care should be taken for the sake of efficiency, that these IBLDR cards be arranged in order of subroutine appearance on the tape.) No origin cards are required since linkage is unnecessary. A planetary tape must be mounted on Logical Unit 8 and of course the Patched Conic IEDIT tape must be mounted if an IBLDR deck is used.

12 BINARY DECKS

<u>MC13 Deck Name</u>	<u>Routine Name</u>
BM	MAIN for EBTLB
IB	/INPCOM/BLOCK DATA for EBTLB
KB	BEAK
BB	BLDPB
HB	CHNGB
33	CONP33

EH	EHA*
EQ	EQTOR*
NM	NORMP
NP	NUTATE*
BO	OUTB
EB	SBEVB

21 \$IBLDR CARDS

for required decks found on the Patched Conic IEDIT Tape

MF	ADOT/DOT/FNORM
AT	ARKTNS
9B	BIBCD
MC	CROSS
ER	ERROUT
MT	MTRN/VTRT
SN	SETN
TM	TIME(PACKAGE)
LA	LAYOVR
PI	PUTIN
RL	ROVLY
RY	ROYAL
FM	ANTR
SA	STAT
CR	CRITO
PB	PARAB
QC	QUARTC
SO	SORDR
EL	EVDEL
PG	PCHNG
CS	STATP

* Also included in Mark II: PINT

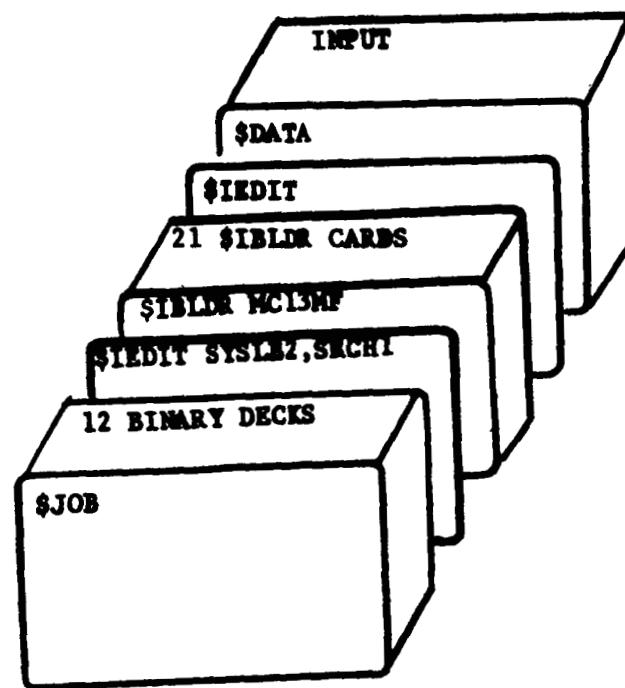


Figure 3-1 EBTLS PROGRAM DECK MAKEUP

SECTION 4

OUTPUT FROM THE EBTLB PROGRAM

This section exhibits print-out from a run made with the EBTLB program simulating the tracking of the Surveyor vehicle from three Earth-based stations, Goldstone, Johannesburg and Woomera.

Figure 4-1 shows the information which is printed at the beginning of each case. Much of this information is printed to provide case identification or to tell the user what he really told the program to do. Overlay input is seen first and consists of the information read into the C-array by input cards for this case. The case header is next seen in a single line of output. "Flight time, 300 days from the starting date, which is seen to be 11:17 pm on June 1, 1966. The beacon's selenographic latitude, longitude and altitude are seen next, followed by the beacon's Cartesian position (first three numbers), velocity (second three numbers) and acceleration relative to the Earth's center, expressed in km, km/sec and km/sec^2 respectively in the true equator and equinox of date coordinate system. The next six rows of numbers headed "BEAK check, etc." is debugging print-out and of little interest to the user. The "P MATRIX INPUT" printed out is the upper half of the (symmetric) state covariance matrix. It shows that the input variance of each component of beacon location error in the north, east, down system is 10 km^2 and that the initial errors are assumed uncorrelated. The control times and station measurements print-outs have been described fully in Reference 1 and will not, therefore, be described here. The normalized P matrix is printed out in coordinates of the beacon's local tangent plane system. "Normalized" means that the off-diagonal elements are correlation coefficients and the diagonal elements are standard deviations (square roots of the corresponding elements). The three columns of zeros following the normalized P-matrix are the initial correlations between the beacon's location errors and the station biases in effect for this case. Reference 1 describes the code for identifying these biases. The standard deviations of the biases considered form the last block of data in Figure 4-1.

Figure 4-2 shows the critical event schedule of station in-view and out-of-view times. Since the beacon is at selenographic longitude $-43^{\circ}32'$, it is never occulted from the Earth by the Moon.

Figure 4-3 shows a portion of the normal print-out seen during execution of the program followed by end conditions for the case. The blocks occur at regular output times (Event 1) critical times of in-view or out-of-view (Event 2) and at the end of the case(Event 7). The numbers labelled X, Y, and Z are the beacon's position components (km) and XD, YD, ZD are the beacon's velocity components (km/sec) relative to the Earth's center expressed in true equator and equinox of date coordinates. The current upper left 3x3 portion of the covariance matrix is printed out each time in north-east-down coordinates, showing the accuracy to which the beacon's location is known, statistically speaking. Finally, the normalized P-matrix and the correlation between beacon location and station biases is printed. Standard deviations of the station biases are seen to be unchanged for this case because these biases were considered but not solved for in this case.

OVERLAY INPUT
 102 0.66060000E 04 103 0.23170000E 04 -0 -0.
 106 -0.24800000E 01 107 -0.43319999E 02 108 0.
 113 0.09000000E 02 114 0.09999999E 02 115 0.09999999E 02 -0 -0.

WDL-TR3007

STARTING CONDITIONS FOR CASE 1 ROOT,RA,DEC. FROM 3 STATIONS TRACK SURVEYOR NO RA-DEC BIAS

FLIGHT TIME 30000.0000DH.MS
 REF. TIME 0. DH.MS
 DATE 6606.01 FOATE 2317.0000
 BEACON COORDINATES
 SELENGRAPHIC LATITUDE= -2.490000 DEG. LONGITUDE= -43.320000 DEG ALTITUDE= 0. MET
 CARTESIAN; EARTH CENTERED; TRUE EARTH EE DATE
 -0.21975911E 06 -0.28132527E 06 -0.12125844E 06 0.81879999E 00 -0.54405137E 00 -0.34011580E-00
 0.16150171E-05 0.20485006E-05 0.88226514E-06

BEAK CHECK AP TRANSPOSE AND TRANPOSE
 0.22277000E-01 -0.37891604E-00 0.92516284E 00
 -0.11026112E-00 0.91881602E 00 0.37897156E-00
 -0.99365288E 00 -0.11045185E-00 -0.21311243E-01
 -0.47902949E-02 0.39917977E-01 0.16464426E-01
 -0.99368248E 00 -0.93885542E-01 -0.61484833E-01
 0.11015702E-00 -0.91794845E 00 -0.37861373E-00

P MATRIX INPUT
 0.09999999E 02 0.
 0.
 0.09000000E 02 0.
 0.
 0.09999999E 02

NEW CONTROL TIMES AT 0 DAY 0 HRS 0 MIN 0.000 SEC
 START 0 DAY 0 HRS 0 MIN 0.000 SEC
 STOP 7 DAY 0 HRS 0 MIN 0.000 SEC
 OINTV 0 DAY 1 HRS 0 MIN 0.000 SEC

*** EARTH-BASED TRACKING IN EFFECT ***

NUMBER	NAME	LOCATION	OBSERVES
1	GOLDSN	LATITUDE = 35.38950 DEG LONGITUDE = 243.15176 DEG ALTITUDE = 1037.53996 MET	AT INTERVALS OF 1800.00 SEC WHEN ELEVATION IS ABOVE 0. DEG BUT LESS THAN 90.00 DEG
		MEASUREMENT ERROR SOURCES	RANDOM BIAS
		RANGE RATE (MET/SEC)	0.00010 0.0183
		RIGHT ASCENSION (MR)	0.20000 0.
		DECLINATION (MR)	0.20000 0.
		STATION LATITUDE (MET-N)	0. 60.0000
		STATION LONG. (MET-EAST)	0. 60.0000
		STATION ALTITUDE (MET-D)	0. 60.0000
2	JOBWRG	LATITUDE = -25.88735 DEG LONGITUDE = 27.66478 DEG ALTITUDE = 1361.91997 MET	AT INTERVALS OF 1800.00 SEC WHEN ELEVATION IS ABOVE 0. DEG BUT LESS THAN 90.00 DEG
		MEASUREMENT ERROR SOURCES	RANDOM BIAS
		RANGE RATE (MET/SEC)	0.00010 0.0183
		RIGHT ASCENSION (MR)	0.20000 0.
		DECLINATION (MR)	0.20000 0.
		STATION LATITUDE (MET-N)	0. 60.0000
		STATION LONG. (MET-EAST)	0. 60.0000
		STATION ALTITUDE (MET-D)	0. 60.0000
3	WONERA	LATITUDE = -31.38287 DEG LONGITUDE = 136.86502 DEG ALTITUDE = 150.79000 MET	AT INTERVALS OF 1800.00 SEC WHEN ELEVATION IS ABOVE 0. DEG BUT LESS THAN 90.00 DEG
		MEASUREMENT ERROR SOURCES	RANDOM BIAS
		RANGE RATE (MET/SEC)	0.00010 0.0183
		RIGHT ASCENSION (MR)	0.20000 0.
		DECLINATION (MR)	0.20000 0.
		STATION LATITUDE (MET-N)	0. 60.0000
		STATION LONG. (MET-EAST)	0. 60.0000
		STATION ALTITUDE (MET-D)	0. 60.0000

THE COVARIANCE MATRIX IS DIMENSIONED 3 X 15

NORMALIZED P MATRIX

	N	E	D
N	0.31622776E 01	0.	0.
E		0.31622776E 01	0.
D			0.31622776E 01
102	0.	0.	0.
107	0.	0.	0.
108	0.	0.	0.
109	0.	0.	0.
202	0.	0.	0.
207	0.	0.	0.
208	0.	0.	0.
209	0.	0.	0.
302	0.	0.	0.
307	0.	0.	0.
308	0.	0.	0.
309	0.	0.	0.

STD. DEV. OF UNSOLVED FOR PARAMETERS

102 0.1830000E-04	107 0.6000000E-01	108 0.6000000E-01
109 0.6000000E-01	202 0.1830000E-04	207 0.6000000E-01
208 0.6000000E-01	209 0.6000000E-01	302 0.1830000E-04
307 0.6000000E-01	308 0.6000000E-01	309 0.6000000E-01

STATION CRITICAL EVENT AND CONDITIONS				RANGE	AZIMUTH	ELEVATION
TIME FROM EPOCH	EVENT	EVENT				
0 DAY 0 MRS 0 MIN 0.000 SEC	GOLDSN OFF	0.37988630E 06	95.000	-27.312		
0 DAY 0 MRS 0 MIN 0.000 SEC	JOBURG ON	0.37166333E 06	-84.771	56.495		
0 DAY 0 MRS 0 MIN 0.000 SEC	WOMERA OFF	0.38017631E 06	-140.534	-30.009		
0 DAY 2 HRS 26 MIN 1.000 SEC	GOLDSN ON	0.377220381E 06	114.567	-0.025		
0 DAY 4 HRS 24 MIN 41.529 SEC	JOBURG OFF	0.37758163E 06	-111.360	0.018		
0 DAY 7 HRS 53 MIN 16.059 SEC	WOMERA ON	0.37808345E 06	113.221	-0.005		
0 DAY 12 HRS 36 MIN 35.277 SEC	GOLDSN OFF	0.37875891E 06	-116.854	0.016		
0 DAY 15 HRS 34 MIN 4.280 SEC	JOBURG ON	0.37922321E 06	113.514	-0.011		
0 DAY 22 HRS 4 MIN 7.137 SEC	WOMERA OFF	0.38022755E 06	-115.933	0.008		
1 DAY 3 HRS 33 MIN 55.316 SEC	GOLDSN ON	0.38108796E 06	119.687	-0.023		
1 DAY 5 HRS 27 MIN 30.953 SEC	JOBURG OFF	0.38140730E 06	-115.751	0.019		
1 DAY 8 HRS 36 MIN 12.794 SEC	WOMERA ON	0.38192217E 06	117.634	-0.014		
1 DAY 13 HRS 18 MIN 24.435 SEC	GOLDSN OFF	0.38268533E 06	-121.246	0.017		
1 DAY 16 HRS 26 MIN 49.498 SEC	JOBURG ON	0.38321841E 06	117.186	-0.013		
1 DAY 23 HRS 8 MIN 1.326 SEC	WOMERA OFF	0.38437510E 06	-119.428	0.013		
2 DAY 4 HRS 38 MIN 56.141 SEC	GOLDSN ON	0.38531244E 06	123.023	-0.020		
2 DAY 6 HRS 28 MIN 51.857 SEC	JOBURG OFF	0.38564616E 06	-118.522	0.020		
2 DAY 9 HRS 24 MIN 32.593 SEC	WOMERA ON	0.38616320E 06	120.350	-0.016		
2 DAY 14 HRS 6 MIN 29.035 SEC	GOLDSN OFF	0.38696335E 06	-123.780	0.009		
2 DAY 17 HRS 12 MIN 2.287 SEC	JOBURG ON	0.38752836E 06	119.183	-0.010		
3 DAY 0 HRS 8 MIN 2.189 SEC	WOMERA OFF	0.38874575E 06	-121.056	0.014		
2 DAY 6 MRS 28 MIN 51.857 SEC	JOBURG OFF	0.38896919E 06	124.310	-0.020		
3 DAY 5 HRS 38 MIN 25.242 SEC	GOLDSN ON	0.39002164E 06	-119.512	0.020		
3 DAY 7 HRS 26 MIN 41.433 SEC	JOBURG OFF	0.39052294E 06	121.187	-0.015		
3 DAY 10 HRS 17 MIN 47.664 SEC	WOMERA ON	0.39131039E 06	-124.275	0.001		
3 DAY 15 HRS 0 MIN 23.164 SEC	GOLDSN OFF	0.39186047E 06	119.408	-0.006		
3 DAY 18 HRS 6 MIN 42.714 SEC	JOBURG ON	0.39302294E 06	-120.787	0.004		
4 DAY 1 HRS 2 MIN 11.205 SEC	WOMERA OFF	0.39390106E 06	123.555	-0.025		
4 DAY 6 HRS 30 MIN 28.027 SEC	GOLDSN ON	0.39420740E 06	-118.756	0.020		
4 DAY 8 HRS 19 MIN 17.664 SEC	JOBURG OFF	0.39467677E 06	120.169	-0.013		
4 DAY 11 HRS 14 MIN 27.335 SEC	WOMERA ON	0.39539056E 06	-122.798	0.002		
3 DAY 18 HRS 6 MIN 42.714 SEC	GOLDSN OFF	0.39587702E 06	117.947	-0.003		
4 DAY 15 HRS 58 MIN 24.949 SEC	WOMERA ON	0.39686804E 06	-118.780	0.009		
4 DAY 19 HRS 3 MIN 7.894 SEC	JOBURG ON	0.39760591E 06	121.013	-0.023		
5 DAY 1 HRS 49 MIN 24.789 SEC	WOMERA OFF	0.39786956E 06	-116.462	0.010		
5 DAY 7 HRS 14 MIN 42.796 SEC	GOLDSN ON	0.39828075E 06	117.508	-0.012		
5 DAY 9 HRS 5 MIN 58.390 SEC	JOBURG OFF	0.39885815E 06	-119.616	0.002		
5 DAY 12 HRS 12 MIN 30.757 SEC	WOMERA ON	0.39923704E 06	115.036	-0.000		
5 DAY 16 HRS 58 MIN 20.414 SEC	GOLDSN OFF	0.39996339E 06	-115.348	0.020		
5 DAY 19 HRS 59 MIN 30.949 SEC	WOMERA OFF	0.40050169E 06	117.056	-0.007		
6 DAY 7 HRS 52 MIN 2.646 SEC	GOLDSN ON	0.40069976E 06	-112.908	0.008		
6 DAY 9 HRS 46 MIN 53.882 SEC	JOBURG OFF					

Figure 4-2

6 DAY 20 HRS 54 MIN 35.656 SEC

EARTH CENTERED, TRUE EEDATE

CASE 1 REC. 165 EVENT 2

X 0.31445778E 06 Y -0.21215461E 06 Z -0.13211924E 06 XD 0.61413720E 00 YD 0.69006101E 00 ZD 0.29130871E-00
 CURRENT 3X3 P COVARIANCE MATRIX, N-E-D
 0.49757870E 01 0.29760300E-00 -0.44307110E-00
 0.29760300E-00 0.76687352E 01 -0.25015339E 01
 -0.44307110E-00 -0.25015339E 01 0.71615643E 01

6 DAY 21 HRS 0 MIN 0.000 SEC

EARTH CENTERED, TRUE EEDATE

CASE 1 REC. 166 EVENT 1

X 0.31465688E 06 Y -0.21193073E 06 Z -0.13202471E 06 XD 0.61350259E 00 YD 0.69048171E 00 ZD 0.29157175E-00
 CURRENT 3X3 P COVARIANCE MATRIX, N-E-D
 0.49697568E 01 0.29508840E-00 -0.44848394E-00
 0.29508840E-00 0.76625963E 01 -0.25099698E 01
 -0.44848394E-00 -0.25099698E 01 0.71490639E 01

6 DAY 22 HRS 0 MIN 0.000 SEC

EARTH CENTERED, TRUE EEDATE

CASE 1 REC. 167 EVENT 1

X 0.31685277E 06 Y -0.20943662E 06 Z -0.13096981E 06 XD 0.60643441E 00 YD 0.69512009E 00 ZD 0.29447797E-00
 CURRENT 3X3 P COVARIANCE MATRIX, N-E-D
 0.49549865E 01 0.29763664E-00 -0.44999919E-00
 0.29763664E-00 0.76538727E 01 -0.25174739E 01
 -0.44999919E-00 -0.25174739E 01 0.71416384E 01

6 DAY 23 HRS 0 MIN 0.000 SEC

EARTH CENTERED, TRUE EEDATE

CASE 1 REC. 168 EVENT 1

X 0.31902316E 06 Y -0.20692591E 06 Z -0.12990449E 06 XD 0.59832163E 00 YD 0.69970109E 00 ZD 0.29735939E-00
 CURRENT 3X3 P COVARIANCE MATRIX, N-E-D
 0.49402457E 01 0.29995548E-00 -0.45170581E-00
 0.29995548E-00 0.76445770E 01 -0.25255550E 01
 -0.45170581E-00 -0.25255550E 01 0.71336730E 01

7 DAY 0 HRS 0 MIN 0.000 SEC

EARTH CENTERED, TRUE EEDATE

CASE 1 REC. 169 EVENT 1

X 0.32116786E 06 Y -0.20439883E 06 Z -0.12882884E 06 XD 0.59216477E 00 YD 0.70422450E 00 ZD 0.30021585E-00
 CURRENT 3X3 P COVARIANCE MATRIX, N-E-D
 0.49255732E 01 0.30219045E-00 -0.45345569E-00
 0.30219045E-00 0.76350813E 01 -0.25337388E 01
 -0.45345569E-00 -0.25337388E 01 0.71256810E 01

END CONDITIONS

7 DAY 0 HRS 0 MIN 0.000 SEC

EARTH CENTERED, TRUE EEDATE

CASE 1 REC. 169 EVENT 7

X 0.32116786E 06 Y -0.20439883E 06 Z -0.12882884E 06 XD 0.59216477E 00 YD 0.70422450E 00 ZD 0.30021585E-00
 CURRENT 3X3 P COVARIANCE MATRIX, N-E-D
 0.49255732E 01 0.30219045E-00 -0.45345569E-00
 0.30219045E-00 0.76350813E 01 -0.25337388E 01
 -0.45345569E-00 -0.25337388E 01 0.71256810E 01

NORMALIZED P MATRIX

	N	E	D
N	0.22193632E 01	0.49277142E-01	-0.76540849E-01
E		0.27631650E 01	-0.34351194E-00
D			0.26693971E 01
102	-0.16615553E-01	-0.34455940E-01	-0.40174420E-01
107	0.59081877E-01	-0.35695450E-01	-0.33131423E-01
108	-0.42399301E-01	-0.16717776E-01	-0.23735490E-01
109	0.77416641E-01	-0.49214893E-01	-0.46334092E-01
202	0.11452073E-02	-0.32941376E-01	-0.27469236E-01
207	-0.48062549E-01	0.51321223E-01	0.51824752E-01
208	-0.78866208E-01	-0.47843378E-01	-0.63546526E-01
209	0.10934691E-00	-0.10792591E-00	-0.10803899E-00
302	-0.13440885E-01	-0.58812350E-01	-0.62207388E-01
307	-0.58185097E-01	0.61559846E-01	0.63369073E-01
308	-0.85545794E-01	-0.51080373E-01	-0.67418019E-01
309	0.10446733E-00	-0.10259391E-00	-0.10470939E-00

STD. DEV. OF UNSOLVED FOR PARAMETERS

102	0.1830000E-04	107	0.6000000E-01	108	0.6000000E-01
109	0.6000000E-01	202	0.1830000E-04	207	0.6000000E-01
208	0.6000000E-01	209	0.6000000E-01	302	0.1830000E-04
307	0.6000000E-01	308	0.6000000E-01	309	0.6000000E-01

Figure 4-3

APPENDIX A

BRIEF DESCRIPTIONS OF SUBROUTINES USED IN THE EBTLB
PROGRAM

<u>Subroutine Name</u>	<u>Description</u>	<u>Deck Name</u>
ADOT	Computes the angle between two given vectors.	MF
ANTR	Interpolates ephemeris information from tape.	FM
ARKTNS	Computes the arc tangent and assigns quadrant.	AT
BEAK	Computes such time dependent data as inertial cartesian coordinates of a lunar beacon given epoch and selenographic coordinates of the beacon.	KB
BIBCD	Converts a binary integer to BCD.	9B
BLDPB	Builds the expanded 3x3 P covariance matrix for the EBTLB program and outputs the error sources included.	BB
BLOCK DATA	/INPCOM/ Loads data into INPCOM common block for use in the EBTLB program.	IB
CHNGB	Makes earth-based tracking station measurements of a lunar beacon and appropriately updates the expanded 3x3 P covariance matrix peculiar to the EBTLB program.	HB
CONP33	Converts a 6-vector to a symmetric 3x3 matrix and outputs the results as the initial P covariance matrix.	33
CRITO	Outputs station in-view, out-of-view critical events.	CR
CROSS	Computes the vector cross product.	MC
DOT	Computes the vector dot product, (ADOT [*]).	MF

* Refer to the bracketed subroutine for further description; e.g., DOT is described in the ADOT subroutine writeup.

EHA	Computes the Greenwich hour angle and the earth's angular velocity.	EH
EQTOR	Computes the transformation from mean equator and equinox of 1950.0 to mean equator and equinox of date.	EQ
ERROUT	Provides programmed response to anticipated errors.	ER
EVDEL	Orders the critical events chronologically and determines the advancing step size.	EL
FNORM	Computes the magnitude of a vector, (ADOT [*]).	MF
LAYOVR	Reads station data cards and converts to proper units.	LA
MAIN	Executive driver for the ELTB program.	BM
MTRN	Multiplies one or more 3x3 matrices by a given 3x3 matrix.	MT
MTRT	Multiplies one or more 3x3 matrices by the transpose of a given 3x3 matrix, (MTRN [*]).	MT
NORMP	Normalizes and outputs the expanded 3x3 P covariance matrix, retaining the standard deviations on the diagonal.	NM
NUTATE	Computes the transformation from earth's mean equator, equinox to earth's true equator and/or moon's true equator, node.	NP
OUTB	Outputs time, state, and 3x3 covariance matrix of beacon location uncertainties, N-E-D.	BO
PARAB	Fits a parabola through three points.	PB
PCHNG	Updates the expanded covariance matrix at an observation.	PG

PUTIN	Reads in measurement error source keys.	PI
QUARTC	Finds the solutions to the quadratic equation.	QC
ROVLY	Reads program input into INPCOM common storage (C array).	RL
ROYAL	Outputs input random,bias, and timing errors associated with station error sources.	RY
SBEVB	Computes a chronologically ordered array of a lunar beacon in-view, out-of-view times for earth-based tracking stations.	EB
SETN	Sets read and write tape numbers.	SN
SORDR	Sorts an array X in ascending order while preserving the correspondence between array X and array NX.	SO
STAT	Computes cartesian coordinates of a point on the surface of a body and the orthogonal transformation relating cartesian to local tangent plane north- east-down.	SA
STATP	Calculates the partials of earth-based tracking station measurements with respect to the extended state vector.	CS
TFRAC	Updates time in whole and fractional days from epoch.	TM
TIMEC	Converts calendar date and time to whole and fractional days from January 1, 1950.	TM
TIMED	Converts time from (days, hours·minutes, seconds) to seconds.	TM
TIMES	Converts time from seconds to alphanumeric days, hours, minutes and seconds.	TM

WDL-TR3007

VNORM Normalizes a vector and also computes the magnitude, MF
(ADOT^{*}).

VTRN Multiplies one or more 3-vectors by a given 3x3 matrix, MT
(MTRN^{*}).

VTRT Multiplies one or more 3-vectors by the transpose of MT
a given 3x3 matrix, (MTRN^{*}).

APPENDIX B

NEW SUBROUTINE WRITE-UPS

This appendix contains descriptions and listings of those subprograms which are unique to the EBTLB program. Descriptions of the remaining subroutines may be found in References 2 and 3.

WDL-TR3007

Subroutine: MAIN(EBTLB)

Purpose: Executive Driver to direct the logic flow of EBTLB,
earth based tracking of a lunar beacon.

Common storages used: INPCOM/C(700)/WCOM/IW(550),CW(1450)

Subroutines required: BEAK,BLDPB,CHNGB,CONP33,EVDEL,OUTB,NORMP,PUTIN,
ROVLY,SBEVB,SORDR,TFRAC,TIMEC,TIMED,TIMES,/INPCOM/
BLOCK DATA(MC13IB)

MAIN (EBTLB)-1

EBTLB PROGRAM

Discussion

The EBTLB program tracks a given lunar beacon. The beacon is restricted in selenographic longitude to be between \pm 80 degrees. This restriction avoids consideration of moon occultation due to lunar librations. (This libration can cause a variation of \pm 7 degrees in the earth-moon vector).

The EBTLB program answers the question "How well can the location of a lunar beacon be determined from earth-based tracking, assuming the position of the moon relative to the earth and the required coordinate transformations are known ?"

The P covariance matrix is expanded from a 3x3 covariance matrix of the error in the estimate of the beacon's location on the moon (north - east - down).

The EBTLB program was created by making modifications to the Patched Conic Error Propagation program; consequently the basic logic flow is nearly identical. The latter program is described in some detail, see MAIN(P.C.), subroutine descriptions for the Mark II Error Propagation Program. Major differences between the two programs are:

1. The vehicle trajectory subroutine PCON was replaced by subroutine BEAK which computes cartesian coordinates of the lunar beacon.
2. The state vector XIN is cartesian coordinates of a lunar beacon earth true equator and equinox of date rather than of a vehicle, mean equator and equinox of 1950.0.

3. P is expanded from a 3 x 3 covariance matrix of the error in the estimate of the beacon's location on the moon (north-east - down), rather than from a 6 x 6 of the error in the estimate of the vehicle state, (EE50).
4. Propagation of the P matrix in time is not required, thus eliminating the need for subroutine PUPT and other related logic in the EBTLB Program.
5. Vehicle onboard and multiple beacon measurements as well as prediction and guidance options have no application in this program.
6. In any given case, changes in the treatment of error sources results in a reinitialization of the P matrix. Unlike the Patched Conic Program, the EBTLB is incapable of "Shuffling" the P matrix.

```

$IBFTC MC13BM NOREF•M94•NODD•XR3
CMC13BM MAIN PROGRAM FOR EARTH-BASED TRACKING OF LUNAR BEACONS
C SUBROUTINES REQUIRED BY MAIN
CMC13KB BEAK
CMC13BB BLDPB
CMC13IB INPCOM BLOCK DATA MODIFIED
CHNGB MODIFIED CHNG
CONP33
CMC1333
CMC13EL EVDEL
CMC13BO OUTB
CMC13NM NORMP
CMC13P1 PUTIN
CMC13RL ROVLY
CMC13EB SBEVB
CMC13SO SORDR
CMC13TM TIME PACKAGE
CMC13TB OTHER SUBROUTINES REQUIRED TO RUN EBTLB
C EBTLB = EARTH-BASED TRACKING OF A LUNAR BEACON
CMC13FM ANTR1
CMC139B BIBCD
CMC13CR CRITO
CMC13MC CROSS
CMC13MF DOT
CMC13EH EHA
CMC13EQ EQTOR
CMC13ER ERROUT
CMC13LA LAYOVR
CMC13MT MTRN,VTRN,MTRI,VTRI
CMC13NP NUTATE
CMC13PB PARAB
CMC13PG PCHNG
CMC13QC QUATRC
CMC13RY ROYAL
CMC13SN SETN
CMC13SA STAT
CMC13CS STATP TEMPORARY
C INPUT REQUIRED
C CASE HEADER CARD (1SW•BCD)
C ROVLY C REFERENCE TIME FROM START EPÜCH
C SEC0 99 DH•MS (MAXIMUM RUN TIME)
C FLTM 101 DH•MS
C DATE 102 YEAR,MONTH•DAY
C FDATE 103 HOUR,MINUTE•SECOND
C BECDM(1) 106 BEACON SELENOGRAPHIC LATITUDE, DEG
C BECDM(2) 107 BEACON SELENOGRAPHIC LONGITUDE, DEG
C BECDM(3) 108 BEACON SELENOGRAPHIC ALTITUDE METERS
C PI(1) 112 P MATRIX 3X3 TYPE (NO SIGNIFICANCE•)
C PI(2) 113 P((1,1))
C PI(3) 114 P(2,2)
C PI(4) 115 P(3,3)
C PI(5) 116 P((1,2)) = P(2,1)
C PI(6) 117 P((1,3)) = P(3,1)

```

```

C      PI(7)    118          P(2,3) = P(3,2)
C      CONTROLS TIMES CARD          ETLB0520
C      STATION CHANGE CARDS          ETLB0530
C      STATION DATA CARDS          ETLB0540
C      COMMON/INPCOM/C(700)/WCOM/IW(150),CW(1450)          ETLB0550
C      ICAS(3), ISC(12), ISEE(12), ISTIM(50), ISW(b)          ETLB0560
C      KEV(8)          ETLB0570
C      AN(3,3), BCD(11), BECRK(3)          ETLB0580
C      EVNT(8), ONAME(3)          ETLB0590
C      P(900), PD(3,3), PI(7)          ETLB0600
C      S(23,12), SEC(12), STIME(5U)          ETLB0610
C      TFORM(6), TIM(2), TST(3), TWT(3)          ETLB0620
C      XIN(9), AP(3,3), APD(3,3)          ETLB0630
C      TI(2)          ETLB0640
C      EQUIVALENCE (C(5),DIR), (C(7),RSPMSD), (C(106),BECDH)          ETLB0650
C      (C(112),PI), (C(200),S)          ETLB0660
C      EQUIVALENCE (CW(1),BERK), (CW(26),PD), (CW(162),P)          ETLB0670
C      (CW(53),XIN), (CW(1140),TSEC)          ETLB0680
C      EQUIVALENCE (IW(8),ISC), (IW(348),NIN), (IW(349),NOUT)          ETLB0690
C      (IW(350),NRP), (IW(351),NCP), (IW(355),ISEE)          ETLB0700
C      (IW(377),ICAS)          ETLB0710
C      EQUIVALENCE (II(1),TW), (II(2),IF)          ETLB0720
C      DATA CMK//JU1 /          ETLB0730
C      DATA (ONAME(J),J=1,3) /&H START, 6H STOP, 6H OINTV/          ETLB0740
C      FORMAT(6I1,1A6)          ETLB0750
C      FORMAT(IH1,1LX,28HSTARTING CONDITIONS FOR CASE •13•1X,11A6)          ETLB0760
C      701  FORMAT(IH,13H FLIGHT TIME •F12•4•2HDH•MS/ 1H REF. TIME,2X,F12•4,ETLB0770
C      702  15HDH•MS•5H DATE,F8•2,2X,SHFDATE,F10•4/29H BEACON COORDINATES /          ETLB0780
C      226H SELENOGRAPHIC LATITUDE=,F11•6,17H DEG. LONGITUDE=,F12•6,15H          ETLB0790
C      3 DEG ALTITUDE=F11•5, 4H MET/ 5H CARTESIAN, EARTH CENTERED,          ETLB0800
C      4 TRUE EARTH EE DATE / (6E2)•8)          ETLB0810
C      706  FORMAT(13,1E12•8•2(3X,1E12•8))          ETLB0820
C      707  FORMAT(21HUNEW CONTROL TIMES AT,6A6)          ETLB0840
C      708  FORMAT(IH ,1A6, 1X,6A6)          ETLB0850
C      709  FORMAT(IH,53X,14HEND CONDITIONS)          ETLB0860
C      NOUT=6          ETLB0870
C      ICAS = L          ETLB0880
C      CONTINUE          ETLB0890
C      READ( 5,700) ISW,BCD          ETLB0900
C      IF (ISW.EQ.1) GO TO 10U          ETLB0910
C      CALL ROVL(C)          ETLB0920
C      ICAS=ICAS+1          ETLB0930
C      WRITE( 6,701) ICAS(1),BCD          ETLB0940
C      ICAS(2) =1          ETLB0950
C      CALL TIMED (C(99),ISECO)          ETLB0960
C      CALL TIMED(C(101),FLTIM)          ETLB0970
C      CALL TIMEC(C(112),C(113),TWO,TF0)          ETLB0980
C      DO 2 I=1,2          ETLB0990
C      BECRK(1)=BECDM(1)*DIR          ETLB1000
C      BECRK(3)=BECDM(3)*CMK          ETLB1010
C      TW=TWO          ETLB1020
C      TF=TF0          ETLB1030
C      2

```

```

C      CALL BEAK( TWO, TFO, BECRK, XIN, GHAR, GM, AP, APD)
C      WRITE(6,702) C(101),C(99),C(102),C(103),BECDM,XIN
C      WRITE(6,800) AP,APD
800    FORMAT(1H0,42H**BEAK CHECK AP TRANSPUSL, APD TRANSPOSE**/(3E20.0))ETLB100
        TEST=ABS(BECDM(21))
        IF(TEST.GT.80.AND.TEST.LT.280.) GO TO 1002
        TSEC = TSECO
        INIT5 = -1
        CALL COMP33(P1,PD)
        PTIME = FLTIM+9999.
        GTIME = PTIME
        BTIME = PTIME
        OBTIME=PTIME
        CONTINUE
        CALL TIMES(TSEC,D,TFORM)
        READ( 2*706) NCH,TST
        IF(NCH.EQ.111) GO TO 260
        WRITE( 6*707) TFORM
        DO 21 I=1,3
        CALL TIMED(TST(I),TWT(I))
        CALL TIMES(TWT(I),D,TFORM)
        WRITE( 6*708) ONAME(I),TFORM
        IF(TWT(1).LT.TSEC) TWT(1)=TSEC
        IF(TWT(1).GT.TWT(2).OR.TWT(3).EQ.0.) GO TO 1001
        IF(NCH.LE.0) GO TO 22
        CALL PUTIN(INITS)
        CALL BLDPB
        WRITE( 6*798) NRP,NCP
798    FORMAT(1H0,36H THE COVARIANCE MATRIX IS DIMENSIONED,13*2H X,13)
        CALL NORP(1,P,NRP,NCP,<PP>)
        MAX=NCP*NRP
        IF(MAX.GT.999.OR.NCP.GT.103) GO TO 1001
        INIT5 = 0
        CONTINUE
        TSTART = TWT(1)
        IF(NCH.LT.0) TSTART=TWT(2)
        KSTART = 1
        IF(TSTART .GT. TSEC) GO TO 40
        CONTINUE
        KSTART = 2
        TSTART = TWT(2)
        TIM(1) = TSEC
        TIM(2) = TWT(2)
        STIME = FLTIM + 9999.
        IF(ISC.EQ.0) GO TO 41
        CALL SBEVB(TI,TIM,ISC,SCR,ISTIM,STIM,MB)
        IK1=1
        KSMAX = MB
        GO TO 41
        CONTINUE
        STIME=FLTIM + 9999.

```

MAIN(EBTLB)-6

```

41    CONTINUE
      OTIME=TSEC + TWI(3)
      EVNT(1)=OTIME
      EVNT(2)=STIME
      EVNT(3)=BTIME
      EVNT(4)=DOSTIME
      EVNT(5)=GTIME
      EVNT(6)=OPTIME
      EVNT(7)=START
      EVNT(8)=FLTIM
      KCULT=-1
      DO 42 I=1,8
        KEV(I)=I
        CALL SORDR(EVNT,KEV,8)
        KINT=-1
        TSTOP=TSEC
        DEL=864000.
        CALL UUTH(P,NRP,NCP)
        IF(KSTART.EQ.1) GO TO 44
43    CONTINUE
        CALL EVDEL(EVNT,KEV,DEL,KCULT)
44    CONTINUE
        KSTOP=KEV(1)
        TSTOP=EVNT(1)
        KTM=1
        IF(KINT.LT.0) GO TO 46
        TDELT=(TSTOP-TSEC)*KSPMSD
        CALL TFAC(TW,TF,TDELT)
        CALL BEAK(TW,TF,BECK,XIN,GHAR,XIN,GHAR,UMGM,AP,APD)
        TSEC=TSTOP
45    CONTINUE
        GO TO (60,47),KSTART
46    CONTINUE
        IF(ISC.EQ.0) GO TO 60
        CALL CHNGb(XIN,GHAR,NRP,NCP,P,SECRR,UMGM,AP,APD)
        KINT=0
        GO TO (61,62),KTM
47    CONTINUE
        TSTOP=TSTOP+DTL
        IF(TSTOP-TSTOP) 45,62,62
48    CONTINUE
        TSTOP=TSTOP
        KTM=2
        GO TO 45
49    CONTINUE
        ICAS(3) = KSTOP
        GO TO (155,160,250,250,250,250,250),KSTOP
50    CONTINUE
        OUTPUT SECTION
        OTIME = OTIME + TWI(3)
        EVNT(1)=OTIME
        ICAS(2) = ICAS(2) + 1
51    CONTINUE
        CALL OUTB(P,NRP,NCP)
        KINT = -1

```

```

GO TO 43
CONTINUE
C      STATION CRITICAL EVENT SECTION
K = 1$TIM(1$K1)
IF (K) 161,161,162
161      K=-K
1$SEE(K) =-1
GO TO 167
162      I$EE(K) =0
SECR(K)=TSEC
CONTINUE
167      IK1 = IK1+1
IF ((IK1-K$MAX)169,169,168
CONTINUE
TIM = TSEC
TIM(2) = TWI(2)
CALL SB$EV(1$TIM,1$C,STCK,1$LT,1$TIM,1$LT,MB)
K$MAX = MB
IK1 = 1
CONTINUE
EVENT(1) = $TIME(IK1)
GO TO 156
CONTINUE
C      THIS SECTION FOR READING NEW CONTROL TIMES
C      FILLING UP STATION ARRAY K$START=2
IF (INCH•LT•J) GO TO 20
GO TO (23,ZU), K$START
250      CONTINUE
WRITE( 6,709)
CALL OUTB(P,NRP,NCP)
CALL NORMP(P,NRP,NCP,KPP)
GO TO 1
CONTINUE
WRITE( 6,701) ICAS(1),BCD
STOP
1002      WRITE (6,799) BECDM(2)
799      FORMAT(1HU,24X,BH*****/7X,32HBEACON SELENCGRAPHIC LUNIUTDE =,
1F12•6•5H DE•/15X,32HDAU CHOICE DUE TO OCCULTING BY MOON/14X,26H
2**EXECUTION TERMINATED**)
GO TO 1001
END

```

Subroutine:

BEAK

WDL-TR3007

Purpose:

Given the epoch and selenographic coordinates of a lunar beacon, BEAK computes inertial cartesian coordinates of the beacon and other time dependent data required by the EBTLB* program.

Calling Sequence: CALL BEAK(TW,TF,B,XB,GHAR,~~GMGM~~,AP,APD)

Input and Output

I/O	Symbolic Name or Location	Program Dimensions	Math Symbol	Data Dimensions or Units	Definition
I	TW			Days	Date, whole days from 1950.0.
I	TF			Days	Date and time, fractional days from 1950.0.
I	B	3		Radians Radians Km	Selenographic coordinates of lunar beacon B(1) = latitude B(2) = longitude B(3) = altitude
Ø	XB	9		Km Km/Sec Km/Sec ²	Earth centered cartesian coordinates of B, true earth equator-equinox of date TW, TF. X Ẋ ẊẊ
Ø	GHAR			Radians	Greenwich hour angle at TW, TF epoch. (Right ascension of the greenwich meridian measured from true equinox of epoch).
Ø	GMGM	ω'		Rad/Sec	Moon's rotation rate.
Ø	AP	3,3		Kms	These matrices are required by subroutine CHNGB to compute the measurement partials for the EBTLB* program.
Ø	APD	3,3		Kms	

Common Storages required: /INPCM/C(700)

Subroutines required: ANTR,CROSS,DOT,EHA,EQTOR,MTRN,NUTATE,STAT

* EBTLB: Earth-based tracking of a lunar beacon.

BEAK-1

PROGRAM NAME	DIMENSION	DESCRIPTION
A	3,3	Transformation matrix, mean EE50 (earth equator, equinox of 1950.0) to mean EEdate.
AN	3,3	Transformation matrix, mean EE50 to true EEdate = (TN) (A).
AP	3,3	Matrix required for computation of measurement partials. = (TRF) (BS).
APD	3,3	Matrix required for computation of measurement partials = (TRF(I,3)) X(AP).
BRAD	2	(1) equatorial radius of moon. (2) polar radius of moon.
BS	3,3	Orthogonal transformation relating cartesian quantities DUM to moon beacon's local tangent plane quantities North, East, and Down. $= \begin{bmatrix} N_1 & E_1 & D_1 \\ N_2 & E_2 & D_2 \\ N_3 & E_3 & D_3 \end{bmatrix}$
D	3,3	Matrix transformation, moon true equator node to selenographic $= \begin{bmatrix} \cos RA & \sin RA & 0 \\ -\sin RA & \cos RA & 0 \\ 0 & 0 & 1. \end{bmatrix}$

BEAK-2

DTM	3,3	Matrix transformation, mean EEdate to selenographic. = (D) (TM).
DUM	9	Cartesian selenographic coordinates of beacon. $DUM(1-3) = R$ $DUM(4-6) = \dot{R} = \dot{\Omega}_{MG} X R$ $DUM(7-9) = \ddot{R} = \ddot{\Omega}_{MG} X \dot{R}$
$\dot{\Omega}_{MG}$	3	Moon rotation vector, (0, 0, $\dot{\Omega}_{MG}$).
RA	3	Argument of moon's prime meridian, measured in moon equator from ascending node of moon equator on ecliptic, (radians).
RB	9	Moon centered coordinates of beacon, true EEdate = (TRF) (DUM)
		$= (TRF) \begin{pmatrix} R_1 & \dot{R}_1 & \ddot{R}_1 \\ \cdot & \cdot & \cdot \\ R_2 & \dot{R}_2 & \ddot{R}_2 \\ R_3 & \dot{R}_3 & \ddot{R}_3 \end{pmatrix}$
RM	9	Earth centered coordinates of moon, EE50. RM(7-9) = VDUM.
TIME		Days from 1950.0 = TW + TF.
TM	3,3	Transformation matrix, mean EEdate to true moon equator , node.
TN	3,3	Transformation matrix, mean EEdate to true EEdate.

TRF	3,3	Transformation matrix, selenographic to true EEdate $= (\text{TN}) (\text{DTM})^T$.
VDUM	3	Equivalent to RM(7-9), mean EE50 $= -\mu_E \frac{\vec{R}_M}{ \vec{R}_M ^3} - \mu_S \left(\frac{\vec{R}_{SM}}{ \vec{R}_{SM} ^3} + \frac{\vec{R}_{SE}}{ \vec{R}_{SE} ^3} \right)$ where \vec{R}_M = position of Moon wrt Earth \vec{R}_{SE} = position of Earth wrt Sun \vec{R}_{SM} = position of Moon wrt Sun
XB	9	Position, velocity, and acceleration coordinates of beacon with respect to earth, true EEdate $= (\text{AN}) (\text{RM}) + \text{RB}$

BEAK-4

```

$IBFTC MC13KB NORF,M94,NODD,XR3          BEAK0010
CMC13KB SUBROUTINE BEAK COMPUTES COORDINATES OF MOON BEACON    BEAK0020
SUBROUTINE BEAK(TW,TF,B,XB,GHAR,OMGM,AP,APD)    BEAK0030
C SUBROUTINES REQUIRED    BEAK0040
C ANTRI    BEAK0050
C CROSS    BEAK0060
C EHA    BEAK0070
C EQTOR    BEAK0080
C MTRN,VTRT    BEAK0090
C NUTATE    BEAK0100
C STAT    BEAK0110
C INPUT    WHOLE DAYS FROM 1950.0    BEAKU120
C TW    FRACTIONAL DAYS FROM 1950.0    BEAK0130
C TF    BEACON SELENOGRAPHIC LATITUDE, LONGITUDE, (RAD,KM)    BEAKU140
C B    BEAK0150
C OUTPUT    EARTH-CENTERED CARTESIAN INERTIAL COORD. OF BEACON,    BEAK0160
C XB    TRUE EEDATE (XB(7)=VDOT)    BEAK0170
C GHAR    GREENWICH HOUR ANGLE AT TW+TF    BEAK0180
C OMGM    ROTATION RATE OF MOON    BEAK0190
C AP    =TRF*(N,E,D) USED FOR PARTIALS IN CHNGB    BEAK0200
C APD    =TRF*(WX)(N,E,D) USED FOR PARTIALS IN CHNGB    BEAK0210
C MATRIX TRANSFORMATIONS,(EARTH-EQUATOR EQUINOX = EE, DATE = TW,TF)    BEAK0220
C AN    MEAN ET 1950.0 TO TRUE EE DATE    BEAK0230
C AN    MEAN LE 1950.0 TO MEAN LE DATE    BEAKU240
C A    MOON TRUE EQUATOR, NODE TO SELENOGRAPHIC    BEAK0250
C D    MEAN EE DATE TO TRUE MOON EQUATOR, NODE    BEAK0260
C TM    MEAN EE DATE TO TRUE EE DATE    BEAK0270
C TN    MEAN EE DATE TO TRUE EE DATE    BEAK0280
C TRF    SELENOGRAPHIC TO TRUE EEDATE = TN*TM TRANPOSE*D TRANSPOSBEAK0280
C VECTORS    BEAKU290
C BS(1,I) UNIT NORTH    BEAK0290
C BS(2,I) UNIT EAST    BEAKG310
C BS(3,I) UNIT DOWN    BEAKU320
C RM    EARTH CENTERED COORDINATES OF MOON    EEE50    BEAKU330
C RB    MOON CENTERED COORDINATES OF BEACON    EEDATE    BEAKU340
C RA    DEFINITIONS OF OTHER VARIABLES    BEAK0320
C RA    ARGUMENT OF MOON PRIME MERIDIAN MEASURED IN MOON EQUATOR    BEAKU360
C RA    FROM ASCENDION NODE OF MOON EQUATOR ON ECLIPIC    BEAK0370
C OMGE    ROTATION RATE OF EARTH    BEAK0380
C COMMON INP/COM/C(700)    BEAK0400
C DIMENSION    A(3,3), AN(3,3), B(3), BODC(10,8)    BEAK0410
C           BRAD(2), BS(3,3), D(3,3), OMG(3), PU(21)
1,           SC(5), TM(3,3), DT(3,3), TRF(3,3)    BEAK0420
2,           DUM(3), RM(9), XB(9), UC(3)    BEAK0430
3,           TN(3,3), VE(21), AP(3,3), APD(3,3)    BEAK0440
4,           VDUM(3)    BEAK0450
5,           EU(VALENCE (C(11),BODC), (C(21),U), (RM(7),VDUM))    BEAK0460
DATA OMGS,D/ 11*0.,1./    BEAK0470
BRAD(1)=BODC(2,3)    BEAK0480
BRAD(2)=BODC(2,4)    BEAK0490
TIME = TW+TF    BEAKU500
CALL EQTOR(TIME,A,OBL)    BEAK0510

```

```

CALL NUTATE(3,TW,TF,TN,TM,RA,OMGM)
CALL MTRN(TN,A,AN)
CALL EHATW,TF,TN(2,1),GHAR,OMGE)
D(1,1)=COS(RA)
D(2,2)=D(1,1)
D(1,2)=SIN(RA)
D(2,1)=-D(1,2)
CALL MTRN(D,TM,DTM)
DO 1 I=1,3
DO 1 J=1,3
TRF(I,J)=0.
DO 1 K=1,3
TRF(I,J)=TRF(I,J)+TN(I,K)*DTM(J,K)
CALL STAT(B,0,BS,DUM,SC,BRAD)
CALL MTRN(TRF,BS,AP)
DO 3 I=1,3
CALL CROSS1(TRF(1,3),AP(1,1),APD(1,1))
OMG(3)=OMGM
CALL CROSS1 OMG,DUM,DUM(4)
CALL CROSS1 OMG,DUM,DUM(4),DUM(7)
CALL MTRN(TRF,DUM,RB)
CALL ANTR1(TW,TF,0,PO,1,V,E,BRAD(1))
DO 10 J=1,3
KK=3+J
RM(J)=POL(KK)
RM(KK)=VE(KK)
R3=DOT(RM,RM)
R3=-U(1)/(R3*SQRT(R3))
DO 11 I=1,3
DUM(I)=-PO(I+6)
KEY=1
R3=DOT(DUM,DUM)
R3=-U(3)/(R3*SQRT(R3))
DO 13 I=1,3
VDUM(I)=VDUM(I)+R3*DUM(I)
DO 14 I=1,3
DUM(I)=RM(I)+DUM(I)
KEY=2
GO TO 12
CONTINUE
CALL MTRN(AN,RM,XB)
DO 20 I=1,9
XB(I)=XB(I)+RB(I)
RETURN
END

```

Subroutine: BLOCK DATA/INPCOM/

Purpose: To supply data to the program concerned with earth-based tracking of a lunar beacon, EBTLB. This Block Data routine is similar to that for the Patched Conic Program; Table 1 describes the pertinent exceptions.

Common storages used: /INPCOM/C(700)

Subroutines required: None

EBTLB/INPCOM/-1

TABLE 1

LOCATION	NAME	COMPILED VALUE	DEFINITION
10	CON(10)	3600.	Step size for subroutine SBEVB, seconds.
106-108	BECDM		Selenographic location of the lunar beacon
106	BECDM(1)	45.	Latitude, degrees
107	(2)	-30.	Longitude, degrees
108	(3)	100.	Altitude, meters.
112	PI		Data required to define initial 3x3 P covariance matrix, local beacon N-E-D coordinates.
113	PI(2)	100.	P(1,1)
114	PI(3)	100.	P(2,2)
115	PI(4)	100.	P(3,3)
116	PI(5)	0.	P(1,2) = P(2,1)
117	PI(6)	0.	P(1,3) = P(3,1)
118	PI(7)	0.	P(2,3) = P(3,2).

EBTLB/INPCOM/-2

\$1BFTC MC131B NOREF,M94,NODD,XR3
 CMC131B INPCOM BLOCK DATA FOR EBTLB
 BLOCK DATA
 COMMON/INPCOM/C(700),
 DIMENSION BODC(110,8), CON(110), BECOM(3), PI(22),
 EQUIVALENCE ((C(111),BODC), ((C(144),SECO), ((C(100),TARG),
 1, ((C(11),CON), ((C(102),DATE), ((C(103),FDATE),
 2, ((C(1101),FLTM), ((C(104),BCEN), ((C(105),TYPEX),
 3, ((C(106),BECDM), ((C(112),PI))
 C ((C(11),PI)= PI/2 C(2)= PI C(3)= 2*PI
 DATA (CON(11),I=1,10)/1.57079633, 3.14159265, 6.2831853
 C(4)= RTD C(5)= DTR C(6)= SPNED C(7)= RSPMSD C(8)= KM/AU
 1.57*2.9577396*0.017452926*86400.0*1574074E-5*149599E9
 C(9)= SPD OF LIGHT C(10)= SBLVb TIME STEP CONSTANT IN SECONDS LBBDO130
 2*299714.0*3600. / LBBDO140
 C NAMES OF BODIES LBBDO150
 DATA (BUDC(11,1),I=1,10)/BHARTN,4HJUN,3HSUN,9HVENUS,4HRAR5,
 16HSATURN,6HJUPITER,6HBODY 8.0HBODY 9,6HBOO 10/
 C GRAVITY CONSTANTS LBBDO160
 DATA(BODC(11,2),I=1,7)/3.98602*2.4700*1008, 132/1545E12, 324764.0*
 14297.0*7418.0E7, 1.26/1.6E6/
 C SEMI-MAJOR AXIS LBBDO170
 DATA(BUDC(11,3),I=1,7)/6.378.105, 1.03, 0.62000, 3400.0*60400.
 1*71350. / LBBDO180
 C SEMI-MINOR AXIS LBBDO190
 DATA(BUDC(11,4),I=1,7)/6.356.0*30, 1/30, 0.69560.0, 62000, 3400.0*54050.
 1.66600. / LBBDO200
 C SPHERE OF INFLUENCE (KM) LBBDO210
 DATA(BUDC(11,5),I=1,7)/925000.0, 60000.0, 1.E11, 0.160000, 266000.
 1*54468.0*4.8E8/ LBBDO220
 C ROTATION RATES (RAD/SEC) LBBDO230
 DATA(BUDC(11,7),I=1,7)/72921.5*2E-4, *266164452E-5, 0.0
 1, 0.0, *108821.7E-4, *17053347E-3, *11774.1111E-3/
 C MAX STEP SIZE (SEC) LBBDO240
 DATA(BUDC(11,6),I=1,7)/43200.0, 21600., 864000., 43200.
 1*43200., 86400., 172000. / LBBDO250
 C INTERPOLATION INTERVALS LBBDO260
 DATA(BODC(11,8),I=1,7)/7*40.0/
 DATA_SECO,TARG,FLTM,DATE,FDATE,BCEN,TYPEX/
 1, 0.0*2.0*30000., 7502*10, 201*25146, 1.0, 0.0/
 DATA (BECDM(1),I=1,3)/ LBBDO270
 1.45*, -30., 100. / LBBDO280
 DATA (PI(1),I=1*22)*0.0*100.0, 100., 100., 18*0.0/
 DATA (C(1),I=167,700)/534*0.0/ LBBDO290
 END

Subroutine: BLDPB

Purpose: To build the P covariance matrix for the EBTLB* program from the input 3 x 3 and other error sources, to compute and store logic necessary to updating this matrix outside BLDPB, and to write out the error sources included.

BLDPB is an extraction of subroutine BLDP. The EBTLB program has no need for onboard measurements or extensive multi-beacon logic, thus BLDPB excludes it also. Also, BLDPB always initializes the extended covariance matrix; it does not contain BLDP's ability to shuffle consideration of error sources.

Calling Sequence: CALL BLDPB

Common storages used: INPCOM/C(700)/WCOM/IW(550),GW(1450)

Subroutines required: ROYAL

* EBTLB: Earth-based tracking of a lunar beacon.

```

$IBFTC MC13BB NOREF,M94,NODD,XR3
CMC13BB BLDPB LOAD AND OUTPUT ELEMENTS OF BEACON 3X3 EXTENDED
      SUBROUTINE BLDPB
COMMON/INPCOM/C(700)/WCOM/IW(550),CW(1450)
DIMENSION ISC(12), IS(11,12), MCOL(5), S(23,12)
      PD(3,3)
      1,      ITEMP(24), IND(4)*, PUV(100)*, EMP(24)
      2,      MP(100)*, P(900)*, EMP(100)
      3,      NIND(4)*, MP(100)
      4,      VAR(100)*, BODC(10,8)
      5,      DIMENSION EQNA(3,24)
      EQUIVALENCE (C(4),RTD), (C(7),RSPMSD), (C(11),BODC)
      (C(200),S), (C(476),B), (C(567),JB)
      1,      (C(585),EMP), (IW(8),ISC), (IW(20),IBC),
      EQUIVALENCE (IW(26),NIN), (IW(180),IS),
      2,      (IW(176),IND), (IW(180),ISOUT),
      (IW(348),NIN), (IW(349),NOUT),
      3,      (MCOL(1),NRP), (MCOL(2),NCP),
      4,      (MCOL(1),P), (MCOL(2),P)
      EQUIVALENCE (CW(26),PD), (CW(62),PUV),
      DATA RTS/*2U626481E6*/
      DATA (EQNA(1,J),J=1,12) /
118HEASTRO UNIT KM/AU ,
218HEARTH M FRAC/SUN ,
318HMMOON M FRAC/SUN ,
418HVENUS M FRAC/SUN ,
518HMARS M FRAC/SUN ,
618HJUPTR M FRAC/SUN ,
718HSATRN M FRAC/SUN ,
818HMRCRY M FRAC/SUN ,
918H2ND ZONAL HARM ,
118H3RD ZONAL HARM ,
218H4TH ZONAL HARM ,
318H5TH ZONAL HARM ,
/ DATA(EQNA(1,J),J=13,24)/
418H1ST LONG HARM RAD ,
518H2ND LONG HARM RAD ,
618H3RD LONG HARM RAD ,
718H4TH LONG HARM RAD ,
818H1ST LUNAR KM**2 ,
918H2ND LUNAR KM**2 ,
118H3RD LUNAR KM**2 ,
218H1ST DRAG 1/(KM) ,
318H2ND DRAG 1/(KM) ,
418HSOLAR RAD K**3/S*S ,
518HVENT T-R KM/SEC ,
618HSPEED LGT ERROR ,
DO 10 I=1,4
10 NIND(1)=0
      NPRNT=1
      NTIME=1
      N=0
      BDBB0010
      BDBB0020
      BDBB0030
      BDBB0040
      BDBB0050
      BDBB0060
      BDBB0070
      BDBB0080
      BDBB0090
      BDBB0100
      BDBB0110
      BDBB0120
      BDBB0130
      BDBB0140
      BDBB0150
      BDBB0160
      BDBB0170
      BDBB0180
      BDBB0190
      BDBB0200
      BDBB0210
      BDBB0220
      BDBB0230
      BDBB0240
      BDBB0250
      BDBB0260
      BDBB0270
      BDBB0280
      BDBB0290
      BDBB0300
      BDBB0310
      BDBB0320
      BDBB0330
      BDBB0340
      BDBB0350
      BDBB0360
      BDBB0370
      BDBB0380
      BDBB0390
      BDBB0400
      BDBB0410
      BDBB0420
      BDBB0430
      BDBB0440
      BDBB0450
      BDBB0460
      BDBB0470
      BDBB0480
      BDBB0490
      BDBB0500
      BDBB0510

```

```

DO 16 I=1,24
IF(IEMP(1))17,16,16
C HERE WE ARE IF THE ITH EQ OF MO PARAM IS TO BE SOLVED FOR
MP(N)=I
VAR(N)=EMP(1)
IEMP(1)=-N-3
16 CONTINUE
NIND(1)=N
NN=N
IF(1ISC) 30,30,20
20 CONTINUE
GO TO (201,202),NPRNT
201 WRITE(NUOUT,1U1)
101 FORMAT(1HU,4X,38H*** EARTH-BASED TRACKING IN EFFECT ***/7H NUMBERDBB0660
1,6X,4HNAME,22X,8HLOCATION,42X,8HOBSEVES)
202 DO 28 I=1,12
ISTAT=ISC(1)
IF(1STAT)30,30,21
21 CONTINUE
GO TO (211,212),NPRNT
211 SLAT=S(3,1STAT)*RTD
SLON=S(4,1STAT)*RTD
AH=S(5,1STAT)*RTD
ALT=S(6,1STAT)*1000.
EM=S(7,1STAT)*RTD
WRITE(NUOUT,1U2) ISTAT,S(1,1STAT),SLAT,S(2,1STAT),SLON,AH,SALT,EM
102 FORMAT(1HU,2X,12.8XA6,12X,11HLATITUDE =,F10.5,4H DEG,20X,15HAT
1N INTERVALS GF,F9.2,4H SEC/31X,11HLONGITUDE =,F10.5,4H DEG,20X,23HWHEDBB0800
2N ELEVATION IS ABOVE F7.2,4H DEG/31X,11HALITUDE
3 X,13HBT LESS THAN F7.2,4H DEG)
WRITE(NUOUT,1U3)
103 FORMAT(1HU,28X,29HMEASUREMENT ERROR SOURCES,11X,6HRANDOM,9X,4HBLIASDBB0840
1)
212 DO 27 J=2,11
GO TO (22,24),NTIME
22 IF((IS(J,1STAT)=EQ.U) GO TO 27
CALL ROYAL(1,I,S(1,1STAT),J,S(1,1STAT),NOUT)
IF((IS(J,1STAT)=GT.0) GO TO 27
23 NN=NN+1
IS(J,1STAT)=-NN-3
GO TO 26
24 IF((IS(J,1STAT)=LE.0.OR.IS(J,1STAT)=EQ.2) GO TO 27
NN=NN+1
IS(J,1STAT)=NN+3
25 MP(NN)=100*ISTAT+J-1
JK=J+12
VAR(NN)=S(JK,1STAT)
27 CONTINUE
28 CONTINUE
30 CONTINUE
IBC=0
I=2*NTIME

```

```

NIND(1)=NN-N
N=NN
GO TO 50,60,NTIME
DO 59 I=1,24
IF (IEMP(1))59,59,51
51 NN=NN+1
MP(NN)=1
VAR(NN)=EMP(I)
TEMP(I)=NN+3
CONTINUE
NIND(3)=NN-N
N=NN
NTIME=2
NPRNT=2
GO TO 2U
60 CONTINUE
NK=NIND(1)
NKL=NK+NIND(2)
NKLM=NKL+NIND(3)
NKLN=NKLM+NIND(4)
NRP=3+NKL
NCP=3+NKLMN
MCOL(3)=3+NK
MCOL(4)=NRP+1
MCOL(5)=2+NKLM
IF (IEMP(24)) 55,57,56
MCUL(3)=MCOL(3)-1
GO TO 57
56 MCOL(5)=MCOL(5)-1
CONTINUE
DO 501 I=1,900
P(I)=0.
501 K=0
DO 503 I=1,3
DO 502 J=1,3
K=K+1
P(K)=PD(J,I)
K=I*NRP
502
CONTINUE
IF (NKL.EQ.0) GO TO 505
K=NRP*3+3
DO 504 I=1,NKL
K=K+1
P(K)=VAR(I)
K=K+NRP
504
CONTINUE
IF (NRP.EQ.NCP) GO TO 507
K=NIND(3)+NIND(4)
DO 506 I=1,K
KK=NKL+I
PUV(I)=VAR(KK)
CONTINUE
DO 508 I=1,100

```

```

BDBB1050
BDBB1060
BDBB1070
BDBB1080
BDBB1090
BDBB1100
BDBB1110
BDBB1120
BDBB1130
BDBB1140
BDBB1150
BDBB1160
BDBB1170
BDBB1180
BDBB1190
BDBB1200
BDBB1210
BDBB1220
BDBB1230
BDBB1240
BDBB1250
BDBB1260
BDBB1270
BDBB1280
BDBB1290
BDBB1300
BDBB1310
BDBB1320
BDBB1330
BDBB1340
BDBB1350
BDBB1360
BDBB1370
BDBB1380
BDBB1390
BDBB1400
BDBB1410
BDBB1420
BDBB1430
BDBB1440
BDBB1450
BDBB1460
BDBB1470
BDBB1480
BDBB1490
BDBB1500
BDBB1510
BDBB1520
BDBB1530
BDBB1540
BDBB1550
BDBB1560
BDBB1570

```

508 MPO(I)=MP(I)
DO 509 I=1•4
IND(I)=NIND(I)
K=0
DO 590 I=1,24
IF(IEMP(I)) 585,590,585
IF(K•GT•0) GO TO 586
K=1
WRITE (NOUT,111)
WRITE (NOUT,112) (EQNA(J,I),J=1,3)*EMP(I)
CONTINUE
586 FORMAT(1H0,42X,35H***OTHER DET. UNKNOWNS IN EFFECT***)
111 FORMAT(1H ,3A6,2X,E15•8)
112 RETURN
END

BDBB1580
BDBB1590
BDBB1600
BDBB1610
BDBB1620
BDBB1630
BDBB1640
BDBB1650
BDBB1660
BDBB1670
BDBB1680
BDBB1690
BDBB1700
BDBB1710
BDBB1720

Subroutine: CHNGB

Purpose: To make station observations of a lunar beacon and appropriately update the P covariance matrix. CHNGB is subroutine CHNG modified to compute partials of measurements with respect to beacon location errors (rather than vehicle state) for the EBTLB* program.

Calling Sequence: CALL CHNGB(X,GHA,NRW1,NCOL,P,SECR,OMGM,AP,APD)

Input and Output

I/O	Symbolic Name or Location	Program Dimensions	Math Symbol	Data Dimensions or Units	Definition
I	X	9	XB	Km ; Sec	Position, velocity, acceleration of beacon with respect to Earth, true EDate.
I	GHA			Radians	Greenwich hour angle.
I	NRW1				Number of rows of P.
I	NCOL				Number of columns of P.
I/O	P	NRW1,NCOL			P covariance matrix.
I/O	SECR	12		Seconds	Array of next time to make an observation for each station. If an observation is made in CHNGB at SECR(II), SECR(II) is set to the next time to observe from station ISC(II).
I	OMGM		w		Rotation rate of the moon.
I	AP	3,3	$\frac{\partial S}{\partial \Delta_B}$		Partial of range vector with respect to beacon location errors N,E,D.
I	APD	3,3	$\frac{1}{ w } \frac{\partial \dot{S}}{\partial \Delta_B}$		$\frac{1}{ w } \cdot$ partial of range rate vector with respect to beacon location error N,E,D.

Common storages used: /INPCOM/C(700)/WCOM/IW(550),CW(1450)

Subroutines required: STATP, VTRT, PCHNG

*EBTLB: Earth-based tracking of a lunar beacon.

CHNGB-1

Discussion:

Given $X(1-6)$, coordinates of the lunar beacon with respect to the Earth, True EEdate, and $X(7-9)$ or VDOT, subroutine STATP computes H measurement partials as required. These partials are in the form required by the EBTLB program with two exceptions: STATP computes $H(1-3) = \frac{\partial M}{\partial x}$, and for rate measurements

$H(1-3) = \frac{\partial M}{\partial x}$ and $H(4-7) = \frac{\partial M}{\partial \dot{x}}$. CHNGB utilizes these H partials to compute those required by EBTLB, $HH(1-3) = \frac{\partial M}{\partial \Delta_B}$ or $\frac{\partial \dot{M}}{\partial \Delta_B}$.

Derivation of HH(1-3)

Let M represent the measurement,

Δ_B represent the beacon location errors, n, e, d,

S represent the range vector

$$= X_m - RT + (TRF) (RB)$$

\dot{S} represent the range rate vector.

$$= \dot{X}_m - \Omega \times RT + (TRF) (\omega \times RB)$$

where RT is inertial position of tracking station, true EEdate.

X_m are earth-centered moon coordinates, true EEdate

Ω earth rotation vector $(0, 0, |\Omega|)$

ω , moon rotation vector $(0, 0, |\omega|)$

RB, selenographic coordinates of lunar beacon

TRF, transformation matrix, selenographic to true EEdate.

$HH(1-3) = \frac{\partial M}{\partial \Delta_B}$ or $\frac{\partial \dot{M}}{\partial \Delta_B}$ is the measurement partial required by subroutine

PCHNG to update the expanded 3×3 P covariance matrix for the EBTLB program.

CHNGB-2

$$\frac{\partial M}{\partial \Delta_B} = \frac{\partial M}{\partial S} \cdot \frac{\partial S}{\partial \Delta_B}$$

$$\frac{\partial \dot{M}}{\partial \Delta_B} = \frac{\partial \dot{M}}{\partial S} \frac{\partial S}{\partial \Delta_B} + \frac{\partial M}{\partial S} \frac{\partial \dot{S}}{\partial \Delta_B}$$

$$\text{STATP provides } H(1-3) = \frac{\partial M}{\partial X} = \frac{\partial M}{\partial S}$$

or, in the case of rate measurements

$$H(1-3) = \frac{\partial \dot{M}}{\partial X} = \frac{\partial \dot{M}}{\partial S}$$

$$H(4-6) = \frac{\partial \dot{M}}{\partial X} = \frac{\partial \dot{M}}{\partial S}$$

Subroutine BEAK provides the CHNGB call list with AP and APD

$$AP = \frac{\partial S}{\partial \Delta_B} = TRF \cdot (N, E, D)$$

$$|\omega| (APD) = \frac{\partial \dot{S}}{\partial \Delta_B} = TRF(\omega X) (N, E, D) \\ = (TRF(\omega)) X (TRF \cdot (N, E, D))$$

But $\omega = (0, 0 | \omega |)$,

so $(TRF(\omega)) = |\omega|$ (column 3 of TRF).

$$\text{thus, } \frac{\partial \dot{S}}{\partial \Delta_B} = |\omega| \left(TRF(1, 3) \right) AP = |\omega| (APD)$$

Consequently $\frac{\partial M}{\partial \Delta_B} = (AP)^T (H)$

and $\frac{\partial \dot{M}}{\partial \Delta_B} = (AP)^T (H_{1-3}) + |\omega| (APD)^T (H_{4-6})$

```

$IBFTC MC13HB NOREF,M94,NODD,XR3
CMC13HB CHNGB CALLS STATP AND COMPUTES MEAS PART WRT BEACON LOCA 3X3
SUBROUTINE CHNGB(X,GHA,NRW1,NCOL,P,SECR,OMGM,AP,APD)
C X IS POS, VEL, ACC, OF BEACON WRT CENTER OF EARTH TRUE ET DATE
C GHA IS THE GHA AT TIME OF X
C SUBROUTINES REQUIRED ARE
C VTRN, VTRT (MC13MT)
C STATP STAT,DOT,CROSS
C PCHNG
C

C COMMON/INPCOM/C(700)/WCOM/IW(550),CW(1450)
C DIMENSION ISC(12), ISEE(12), IS(11,12),
C          LL(2),      BODC(10,8),
C          MTYPE(5),   H(12),
C          NCG(6),    SECR(12),
C          BRAD(3),
C          X(9),     AP(3,3),   W(100)
C          EQUIVALENCE (MEAS,MTYPE), (MTYPE(5),ISAN),
C          (LL(1),KJ),
C          (LL(2),JJ),
C          (C(11),BODC)
C          (IW(8),ISC),
C          (IW(53),ISL),
C          (IW(355),ISEE), (CW(52),W),
C          (IW(180),IS),
C          (CW(1140),TIM)
C
C          BRAD(1)=BODC(1,3)
C          BRAD(2)=BODC(1,4)
C          BRAD(3)=BODC(1,7)
C          DO 1000 I=1,3
C          NCH(I)=1
C          MTYPE(4)=-1
C          IF((ISL.NE.0) MTYPE(4)=0
C          NTBIAS=1
C          II=1
C          CONTINUE
C          IF((ISC(II)) 100,100,3
C          IF((ISEE(II))21,4,4
C          IF((SECR(II).GT.TIM) GO TO 21
C          NST=ISC(II)
C          SECR(II)=SECR(II)+S(2,NST)
C          MTYPE(3)=-1
C          MTYPE(2)=-1
C          DO 7 I=8,1U
C          IF((IS(I,NST).NE.0) MTYPE(2)=0
C          ISAN=IS(1,NST)
C          NN=7
C          IF((ISAN.EQ.0) NN=3
C          IK=0
C          DO 20 J=2,NN
C          M=1$J,NST)
C          IF(M.EQ.0) GO TO 20
C          MEAS=J-1
C          MM=MEAS
C          Q=S(MEAS+7*NST)
C
C          S(23,12)CHGB0110
C          P(NRW1,NCOL)CHGB0120
C          HH(J) CHGB0130
C          NCH(9) CHGB0140
C          CHGB0150
C          CHGB0160
C          CHGB0170
C          CHGB0180
C          CHGB0190
C          CHGB0200
C          CHGB0210
C          CHGB0220
C          CHGB0230
C          CHGB0240
C          CHGB0250
C          CHGB0260
C          CHGB0270
C          CHGB0280
C          CHGB0290
C          CHGB0300
C          CHGB0310
C          CHGB0320
C          CHGB0330
C          CHGB0340
C          CHGB0350
C          CHGB0360
C          CHGB0370
C          CHGB0380
C          CHGB0390
C          CHGB0400
C          CHGB0410
C          CHGB0420
C          CHGB0430
C          CHGB0440
C          CHGB0450
C          CHGB0460
C          CHGB0470
C          CHGB0480
C          CHGB0490
C          CHGB0500
C          CHGB0510
C          CHGB0520

```

```

9      CALL STATP(S(3,NST),X   ,GHA,H,IK,MTYPE,BRAD,X(7),Q)
      KJ=3
      CALL VRTI(AP,H,HH)
      GO TO (12,1,12,12,11,11),MM
      DO 10 I=1,3
      HH(I)=HH(I)+OMGM*DOT(H(4),APD(1,I))
10      JJ=U
      IF (M=2) 13,15,14
12      CONTINUE
      KJ=KJ+1
      HH(KJ)=1.
      NCH(KJ)=-M
      GO TO 15
13      JJ=JJ+1
      G(JJ)=1.
      NCG(JJ)=M
      CONTINUE
15      KK=8
      IF (MTYPE(2)*LT.0) KK=11
      DO 18 I=KK,12
      IF (I.EQ.12) GO TO 158
      M=IS(1,NST)
      GO TO 159
158      M=ISL
      IF (MM.NE.1) GO TO 18
      IF (M) 16,18,17
159      CONTINUE
      KJ=KJ+1
      NCH(KJ)=-M
      HH(KJ)=H(I)
      GO TO 18
16      JJ=JJ+1
      NCG(JJ)=M
      G(JJ)=H(I)
      CONTINUE
18      CALL PCHNG(P,NRW1,NCOL,LL,W,G,NCG,HH,NCH,Q)
      20      CONTINUE
      21      CONTINUE
      II=II+1
      IF (II.LE.12) GO TO 1
      RETURN
100     END

```

Subroutine: CONP33

Purpose: To convert a 6-vector to a symmetric 3×3 and output the result so as to describe the initial P covariance matrix.

Calling Sequence:

CALL CONP33 (PI, PD)

Input and Output

I/ \emptyset	Symbolic Name or Location	Program Dimensions	Math Symbol	Data Dimensions or Units	Definition
I	PI	7			Input 6-vector PI(2) - PI(7) PI(1) is ignored.
\emptyset	PD	3,3			Output 3x3 matrix such that $PD(1,1) = PI(2)$ $PD(2,2) = PI(3)$ $PD(3,3) = PI(4)$ $PD(1,2) = PD(2,1) = PI(5)$ $PD(1,3) = PD(3,1) = PI(6)$ $PD(2,3) = PD(3,2) = PI(7)$

Common storages used: None

Subroutines required: None

CONP33-1

```

$IBFTC MC1333 NOREF,M94,NODD,XR3 CP33U010
CMC1333 CONP33 LOADS P1(2 TO 7) INTO APPROP PD(3,3)
      SUBROUTINE CONP33(PI,PDI) CP33U020
      DIMENSION PI(7), PDI(3,3) CP33U030
      DO 1 I=1,3 CP33U040
      DO 1 I=1,2 CP33U050
      DO 1 I=1,1 CP33U060
      1 PI(I,1)=PI(I+1) CP33U070
      K=4 CP33U080
      DO 2 I=1,2 CP33U090
      KK=I+1 CP33U100
      DO 2 J=KK,3 CP33U110
      K=K+1 CP33U120
      PD(I,J)=PI(K) CP33U130
      PD(J,I)=PI(K) CP33U140
      2 WRITE (6,15U) ((PD(I,J),J=1,3),I=1,3)
      15 FORMAT(1H0,2H0,1H0 P MATRIX INPUT/3U,3T2U,8/5U,X,2E2U,8/7U,X,E2U,8) CP33U150
      RETURN CP33U160

```

Subroutine: NORMP

Purpose: To output the normalized P covariance matrix. NORMP is similar to LOUT with two exceptions: 1. the basic P matrix is a 3 by 3 rather than a 6 by 6, as required by the EBTLB^{*} program, and 2. the diagonal terms contain the standard deviations rather than 1.0. .

Calling Sequence: CALL NORMP(P, NR, NC, KPP)

Input and Output

I/O	Symbolic Name or Location	Program Dimensions	Math Symbol	Data Dimensions or Units	Definition
I	P	NR, NC			Covariance matrix.
I	NR				Number of rows of P.
I	NC				Number of columns of P.
0	KPP				KPP = -1, normal KPP = 1, negative element on diagonal of P matrix. KPP = 2, off diagonal of normalized P greater than 1.

Common storages used: /WCOM/IW(550),CW(1450)

Subroutines required: None

*EBTLB stands for Earth-based tracking of a lunar beacon.

```

$IBFTC MC13NM NOREF,M94,NODD,XR3
CMC13NM NORMP OUTPUTS NORMALIZED EXTENDED P MATRIX
SUBROUTINE NORMP(P,NR,NC,KPP)
COMMON/INPCOM/C(700)I/WCOM/IW(155),CW(1450)
DIMENSION P(NR,NC), PUV(IW), MPO(IW),
          B(3), PD(3,3), BN(3)
EQUIVALENCE (IW(76),MPU), (CW(26),PD), (CW( 62),PUV)
DATA (BN(I),I=1,3)/2H N,2H E,2H D/
DATA (BN(I),I=1,3)/2H N,2H E,2H D/
KPP = -1
      WRITE ( 6,701) BN
      FORMAT(1H0,49X,20H NORMALIZED P MATRIX/IH * 39X,A2*2(17X,A2))
1      DO 2 I=1,3
        IF(P(I,I).LT.0.) KPP=1
        D(I)=SQRT(P(I,I))
        PD(I,I) = D(I)
2      DO 3 I=2,3
        K=I-1
3      DO 3 J=1,3
        PD(K,J)=P(K,J)/(D(K)*D(J))
        IF (ABS(PD(K,J)).GT.1.0) KPP=2
        WRITE ( 6,702) (BN(I),I=1,3),J=1,3
        FORMAT(1H * 27X,A2,3T19.8/IH * 27X,A2,19X,2T19.8/IH * 27X,A2,30X,t19.
18) 
        IF (NRE.EQ.3) GO TO 5
        DO 4 I=4,NR
          IF(P(I,I).LT.0.) KPP = 1
          D(I) = SQRT(P(I,I))
          CONTINUE
          IF (NRE.EQ.NC) GO TO 7
          KK = NR+1
          DO 6 I=KK,NC
            JJ = I-NR
            D(I) = SQRT(PUV(IJ))
            CONTINUE
            IF (NRE.EQ.3) GO TO 2
            DO 9 I=4,NC
              KQ=I-3
              DO 6 J=1,3
                B(J)=P(J,I)/(D(J)*D(I))
                IF (ABS(B(J)).GT.1.0) KPP=2
                CONTINUE
                WRITE ( 6,703) MPO(KQ),B
                FORMAT(1H * 25X,14,3E19.8)
                CONTINUE
                K=3
                IF (NRE.LE.K) GO TO 2
                KK=NR-K
                JJ=j
                IF ((KK.LT.3) .AND. (JJ>KK))
                  DO 12 I=1,JJ
12              L=K+1
                DO 11 J=1,1
11              NRMPP0010
NRMPP0011
NRMPP0012
NRMPP0013
NRMPP0014
NRMPP0015
NRMPP0016
NRMPP0017
NRMPP0018
NRMPP0019
NRMPP0020
NRMPP0021
NRMPP0022
NRMPP0023
NRMPP0024
NRMPP0025
NRMPP0026
NRMPP0027
NRMPP0028
NRMPP0029
NRMPP0030
NRMPP0031
NRMPP0032
NRMPP0033
NRMPP0034
NRMPP0035
NRMPP0036
NRMPP0037
NRMPP0038
NRMPP0039
NRMPP0040
NRMPP0041
NRMPP0042
NRMPP0043
NRMPP0044
NRMPP0045
NRMPP0046
NRMPP0047
NRMPP0048
NRMPP0049
NRMPP0050
NRMPP0051
NRMPP0052
NRMPP0053
NRMPP0054
NRMPP0055
NRMPP0056
NRMPP0057
NRMPP0058
NRMPP0059
NRMPP0060

```

```

NRMF U061
NRMF U062
NRMF U063
NRMF U064
NRMF U065
NRMF U066
NRMF U067
NRMF U068
NRMF U069
NRMF U070
NRMF U071
NRMF U072
NRMF U073
NRMF U074
NRMF U075
NRMF U076
NRMF U077
NRMF U078
NRMF U079
NRMF U080
NRMF U081
NRMF U082
NRMF U083
NRMF U084
NRMF U085
NRMF U086
NRMF U087
NRMF U088
NRMF U089
NRMF U090
NRMF U091
NRMF U092
NRMF U093
NRMF U094
NRMF U095
NRMF U096
NRMF U097
NRMF U098
NRMF U099
NRMF U100
NRMF U101
NRMF U102
NRMF U103
NRMF U104
NRMF U105
NRMF U106
NRMF U107
NRMF U108

```

M=K+J
PD(I,J)=P(L,M)/(D(L)*D(M))
IF(ABS(PD(I,J))>1.0000001) KPP=2
CONTINUE
11 PD(I,I)=D(L)
CONTINUE
12 KK=L-J+1
KK=K-K-3
LJ=L-3
WRITE(6,704) (MPO(I),I=KK,LJ)
FORMAT(1H0,4YX,2UDETERMINISTIC EKKURS/1H ,3oX,14,2(1CX,14))
704 KK=KK-1
DO 15 I=1,JJ
KJ=KK+I
KKJ=KJ-3
WRITE(6,703) MPO(KKJ),(PD(I,J),J=1,I)
IF(INC.EQ.0.KJ) GO TO 20
KJ=KJ+1
DO 17 I=KJ,NC
KI=I-3
DO 16 J=1,JJ
L=K+J
B(J)=P(L,I)/(D(L)*D(I))
B(J)=P(L,I)/(D(L)*D(I))
IF(ABS(B(J))>GT,1.) KPP=2
CONTINUE
16 WRITE(6,703) MPO(KQ),(B(J),J=1,I)
CONTINUE
17 K=K+JJ
GO TO 1
CONTINUE
18 IF(NR.EQ.NC) GO TO 22
L=NC-2
M=NC-3
I=0
DO 21 J=L,M
I=I+1
D(J)=SQRT(PUV(I))
21 WRITE(6,705) DEV. OF UNSOLVED FOR PARAMETERS
FORMAT(1H0,42X,36HSTD. DEV. OF UNSOLVED FOR PARAMETERS)
WRITE(6,706) (MPC(I),D(I),I=L,M)
FORMAT(13,EL,7,I4,E15.7,I4,E15.7)
22 IF(KPP.LT.0) GO TO 31
WRITE(6,707) KPP
FORMAT(21HEXECUTION TERMINATED/1H ,11,1X,27H TYPE PROBLEM WITH P
1MATRIX)
CONTINUE
31 RETURN
END

```

$IBFTC MC13BO NOREF,M94,NODD,XR3
CMC13BO OUTB OUTPUT ROUTINE FOR EARTH-BASED TRACK OF LUNAR BEACONS
SUBROUTINE OUTB(P,NR,NC)
COMMON/INPCOM/C(700),WCOW/LW(550),CW(1450)
DIMENSION P(NR,NC), A(6), ICAS(3), BODC(10,0),
1, XIN(6), ARMS(26), MPC(100)
EQUIVALENCE (C(11),BODC)
EQUIVALENCE (LW(76),MPO), (LW(377),ICAS)
1, (CW(53),XIN), (CW(114),TSEC)
DATA (B(I)*I=1,6)/2H X,2H Y,2H Z,ZHAD,ZHYD,2HLD/
700 FORMAT(//1HJ,6A6,5D5,4HCASL,13,2X,4HREC•,15,2X,5HVENT,12)
701 FORMAT(1H •A6,24H CENTERED, TRUE EEDATE )
702 FORMAT(6(2X,A2,1X,E15.8))
703 FORMAT(1H •37X,18HCURRENT RMS VALUES)
704 FORMAT(1H •26X,38HCURRENT 3X3 PCOVARIANCE MATRIX, N-E-D/(32X,2E20)OUTB140
1•8)
705 FORMAT(6(1X,I4,1H=,E14.7))
CALL TIMES(TSEC,D,A)
WRITE (6 ,700) A,ICAS
WRITE (6 ,701) BODC(1,1)
WRITE (6 ,702) (B(I),XIN(I)*I=1,6)
WRITE (6 ,704) ((P(I,J),J=1,3),I=1,3)
K=NR-3
IF (K.EQ.0) GO TO 3
WRITE (6 ,703)
DO 2 I=4, NR
N=I-3
ARMS(N)=SQRT(P(1,1))
2 WRITE (6 ,705) (MPO(I),ARMS(I)*I=1,K)
3 CONTINUE
RETURN
END

```

Subroutine: OUTB

Purpose: To write the output required during execution of the EBTLB program. Output includes earth centered cartesian coordinates of the Lunar beacon, true equator-equinox of date, the 3×3 P covariance matrix of beacon location uncertainties N-E-D, and the square roots of the diagonal elements of solved for parameters. This output is at regular output points and other critical event times.

Calling Sequence: CALL OUTB(P,NR,NC)

Input and Output

I/O	Symbolic Name or Location	Program Dimensions	Math Symbol	Data Dimensions or Units	Definition
I	P	NR ,NC			Covariance matrix.
I	NR				Number of rows of P.
I	NC				Number of columns of P.

Common storages used: INPCDM/C(700)/WCDM/IW(550),CW(1450)

Subroutines required: TIMES

OUTB-1

Subroutine:

SBEVB

Purpose:

SBEVB creates a chronologically ordered array of those times when the lunar beacon comes into view and goes out of view of earth-based tracking stations. This subroutine is specifically designed for the EBTLB* program. It is similar to SBEV with these exceptions: SBEVB considers occulting by earth only, is concerned with earth-based tracking stations only, and looks for one lunar beacon thereby requiring subroutine BEAK rather than the trajectory subroutine PCON.

Calling Sequence: CALL SBEVB(TI,TIM,ISS,SECR,ISEE,ISTIM,STIME,MB)

Input and Output

I/O	Symbolic Name or Location	Program Dimensions	Math Symbol	Data Dimensions or Units	Definition
I	TI	2		Days	Date in whole and fractional days from 1950.0.
I	TIM	2		Seconds	Minimum and maximum times (from epoch) of interest defining interval of investigation.
I	ISS	12			Numbers of stations or beacons to be considered. ISS(J) = 0 terminates list of numbers.
Ø	SECR	12		Seconds	Time (from epoch) to make observations for stations given in corresponding ISS.
Ø	ISEE	12			In view indicator for station given in corresponding ISS. Set -1 out of view, 0 in view.
Ø	ISTIM	50			Event indicator, in conjunction with STIME times. +, station into view -1, station out of view
Ø	STIME	50		Seconds	Ordered array of times (from epoch) at which ISTIM critical events occur.
Ø	MB				KKK = total number of critical events times STIME computed.

Common storages used: INPCOM/C(700)/WCOM/IW(550),CW(1450)

Subroutines required: BEAK,CRITO,PARAB,QUARTC,SORDR,STAT,TFRAC,VTRN,VRT

* EBTLB: Earth-based tracking of a lunar beacon.

SBEVB-1

COMMON

HEAD	LOCATION	PROGRAM NAME	DIMENSION	DESCRIPTION
INPCOM	C(7)	RSPMSD		Reciprocal seconds per mean solar day = 1/86400.
	C(10)	DELTX		SBEVB step size in seconds (set to 3600. or 1 hour by Block Data).
	C(11) to C(90)	BODC(I,J)	10,8	J th body constant for body number I. (Refer to BLOCK DATA/INPCOM/ description).
	C(200) to C(475)	S(I,J)	23,12	I data item for station number J.
WCOM	CW(1) to CW(3)	BECRK	3	Selenographic coordinates of the lunar beacon, latitude and longitude in radians, altitude in Km.

SBEVB-2

```

S1BFTC MCL3EB NUREF *NJD0,AR3
CMC13EB SBEVB FOR EARTH BASED TRACKING OF A LUNAR BEACON
SUBROUTINE SBVB(TI,TIM,ISS,SEC,R,STAT,STIM,STIME,MB)
C
C SUBROUTINES REQUIRED
C      BEAK      CRITC      PARAB      QJARTC      SQRDR      STAT
C      TFRAC      VTRN,VTRT
C
C COMMON/INPCOM/C(700)/WCUM/IW(500),CW(1450)
C      DIMENSION A(2), AP(3,3), BD(3,3),
C      BECRK(3), BODC(10,8),
C      DUM(3), EN(3,3),
C      IMA(12), ISC(12), ISE(12), ISEK(12),
C      LOS(12), ISTIM(50),
C      RS(3), RT(3),
C      SC(2), SEC(12), STIM(50), STIMR(3,12),
C      TDM(3), TI(2), TIM(2),
C      X(9)
C      EQUIVALENCE (C(7),RSPMSD), (C(1),DELTX), (C(11),BODC)
C      ((200),S)
C      (CW(1),BECRK)
C
C EQUIVALENCE
C      TW=TI(1)
C      TF=TI(2)
C      TSECO=TIM(1)
C      NN=-2
C      TSTP=TSECO
C      TSEC=TSECU
C      FLTIME=TIM(2)
C      DELTY=(FLTIME - TSECO)/3.3
C      IF(DELTY.GT.DELTY) DELTY=DELTX
C      DO 3 I=1,12
C      IF(ISS(1).GT.4,4,2
C      ISC(1)=ISS(1)
C      CONTINUE
C      3
C      I=13
C      NKK=I-1
C      DO 17 I=1,NKK
C      IMA(1)=1
C      17
C      BRAD(1)=BODC(1,3)
C      BRAD(2)=BODC(1,4)
C      KKK=0
C      NHEAD=1
C      CONTINUE
C      10
C      DELT=(TSTP-TSEC)*RSPMSD
C      CALL TFRAC(TWF,DELT)
C      CALL BEAK(TW,TF,BECRK,R,GHAN,OMGM,AP,APD)
C      TSEC=TSTP
C      TSTP=TSTP+DELT
C      NN=NN+1
C      TST(3)=TST(2)
C      TST(2)=TST(1)
C      TST(1)=TSEC
C      DQ 80 II=1,NKK

```

```

NST=ISCV(11)
IF(NST) 80,80,21
CONTINUE
CALL STAT(3,NST),GHAN,EN,RT,SC,BRAD)
DO 22 I=1,3
RT(1)=R(1)-RT(1)
CALL VTRTEN,RT,RS)
EL=ATAN(-RS(3)/SQRT(RS(1)*RS(1)+RS(2)*RS(2)))-S(6,NST)
STOR(3,II)=STOR(2,II)
STOR(2,II)=STOR(1,II)
STOR(1,II)=EL
IF(INN) 25,8,40
IF(EL) 29,29,28
CONTINUE
29  ISEK(11)=1
GO TO 30
CONTINUE
30  ISEK(11)=2
IF(EL.GT.S(7,NST)-S(6,NST))GO TO 296
ISEE(11)=0
SEC(11)=TSEC
KSW=2
GO TO 31
IMAX(11)=2
CONTINUE
31  ISEE(11)=-1
KSW=1
CONTINUE
EL=EL+S(6,NST)
CALL CRITOIS(1,S(1,NST),KSW,RS,EL,6,NHEAD)
GO TO 80
CONTINUE
CALL PARABITST,STOR(1,II),AA)
AD=U.
JJ=1SEK(11)
KSAV=IMAX(11)
IF(JJ.EQ.2) AD=S(7,NST)-S(6,NST)
CONTINUE
CALL QUARTC(STOR(2,II),AD,AA,DT,KK)
IF(KK) 42,44,44
CONTINUE
41  IF(JJ+KSAV.EQ.3) GO TO 43
IF(JJ.EQ.3.AND.AA(1).GT.0.) JJ=2
GO TO 79
42  JJ=3
AD=U.
GO TO 41
CONTINUE
44  GO TO (445,446,445),JJ
IF(STOR(2,II).LT.0..AND.KSAV.EQ.1) GO TO 43
CONTINUE
445  DELT=TST(2)-TST(3)
DTEST=DT(2)

```

```

      IF (JJ+KSAV.EQ.4) DTEST=DT(1)
      IF (DTEST) 45,45,79
      CONTINUE
      IF (DELT+DTEST.LT.0.) GO TO 42
      IF (JJ.EQ.2) DSAV=DT(1),
      CONTINUE
      DUM(1)=STOR(2,II)
      DUM(3)=STOR(3,II)
      TDUM(1)=TST(2)- TST(1)
      TDUM(2)=TDUM(1) + DTEST
      TDUM(3)=TST(3)- TST(1)
      NJAZ=1
      CONTINUE
      NJAZ=NJAZ+1
      TDEL=T=TDUM(2)*RSPMSD
      CALL TFRAC(TW,TF,TWX,TFX,TDEL,T)
      CALL BEAK(TWX,TFX,BECK,XX,GHAM,OMGM,AP,APD)
      CALL STAT(S(3,NST),GHAM,EN,RT,SC,BRAD),
      DO 54 I=1,3
      RT(I)=XX(I)-RT(I)
      CALL VTRT(EN,RT,RS)
      EL=ATAN(-RS(3)/SQRT(RS(1)*RS(1)+RS(2)*RS(2)))-S(6,NST)
      CON=.002
      IF (ABS(EL-AD)-CON) 61,56,56
      56  CONTINUE
      IF (NJAZ.GT.10) GO TO 61
      DUM(2)=EL
      CALL PARAB(TDUM,DUM,BB)
      CALL QUARTC(DUM(2),AD,BB,DT,KK)
      IF (KK) 42,57,57
      CONTINUE
      DTEST=DT(2)
      IF (JJ+KSAV.EQ.4) DTEST=DT(1)
      K=1
      IF (DTEST.GT.0.) K=3
      DUM(K)=DUM(2)
      TDUM(K)=TDUM(2)
      TDUM(2)=TDUM(2)+DTEST
      GO TO 49
      61  CONTINUE
      KKK=KKK+1
      STIME(KKK)=TST(1)+TDUM(2)
      GO TO 62,65,66,JJ
      CONTINUE
      62  ISTIM(KKK) = 11
      EL=EL+S16,NST)
      CALL CRITO(STIME(KKK),S(1,NST)*2,RS,EL,6,NHEAD)
      JJ=2
      IF (IA(1).GT.0.) GO TO 79
      AD=S(7,NST)-S(6,NST)
      GO TO 41
      CONTINUE
      GO TO 66,75,KSAV

```

```

66    ISTIM(KKK)=-11
      EL=EL+S(6,NST)
      CALL CRITO(STIME(KKK),S(1,NST),1,RS,EL,6,NHEAD)
      KSAV=2
      IF(TST(2)+DSAV.LE.TDUM(2)+TST(1).OR.TST(2)+DSAV.GT.TST(1))GO TO 79SBVB1630
      IF(DSAV.GT.U.) GO TO 67
      DTEST=DSAV
      GO TO 48
      CONTINUE
      DUM(1)=STOR(1,II)
      DUM(3)=STOR(2,II)
      TDUM(1)=0.
      TDUM(3)=TST(2)-TST(1)
      TDUM(2)=TDUM(3)+DSAV
      GO TO 49
      CONTINUE
67    ISTIM(KKK)= -11
      EL=EL+S(6,NST)
      CALL CRITO(STIME(KKK),S(1,NST),1,RS,EL,6,NHEAD)
      JJ=1
      IF(AA(2)) 79,79,41
      CONTINUE
      KSAV=1
      ISTIM(KKK) = II
      EL=EL+S(6,NST)
      CALL CRITO(STIME(KKK),S(1,NST),2,RS,EL,6,NHEAD)
      JJ=3
      AD=0.
      GO TO 41
      CONTINUE
79    ISEK(II)=JJ
      IMAX(II)=KSAV
      CONTINUE
      IF(NN) 10,10,81
      IF(TST(2).GE.FLTIM) GO TO 82
      IF(KKK=38) 10,83,83
      KKK=KKK+1
      STIME(KKK)=FLTIM + 99999.
      CONTINUE
      MB=KKK
      CALL SORDR(STIME,ISTIM,KKK)
      RETURN
      END

```

REFERENCES

1. "User's Manual for Mark II Error Propagation Program", WDL-TR2758, 15 February 1966.
2. "Subroutine Descriptions and Listings" Mark II Error Propagation Program and Powered Flight Optimization and Error Analysis Programs WDL-TR2757, 15 February 1966. Volume I.
3. "Subroutine Descriptions and Listings" Mark II Error Propagation Program and Powered Flight Optimization and Error Analysis Programs WDL-TR2757, Volume II.
4. "Computer Program Development Study", WDL-TP2002, 16 September 1965.